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DeVlieg

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(54) **CARBON BRAKE WEAR FOR AIRCRAFT**

(56) **References Cited**

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U.S. PATENT DOCUMENTS

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(*) Notice: Subject to any disclaimer, the term of this patent is extended or adjusted under 35 U.S.C. 154(b) by 1186 days.

This patent is subject to a terminal disclaimer.

2,794,609 A	6/1957	Perry
3,520,575 A	7/1970	Steigerwald
3,537,551 A	11/1970	Serra
3,776,333 A	12/1973	Mathauswer
3,926,479 A	12/1975	Bissell et al.
3,948,569 A	4/1976	Gentet
3,993,174 A	11/1976	Williams et al.
4,006,941 A	2/1977	DeVlieg
4,007,970 A	2/1977	Romero
4,008,868 A	2/1977	Berg
4,076,331 A	2/1978	DeVlieg
4,180,223 A	12/1979	Amberg
4,195,803 A	4/1980	Faure
4,205,735 A	6/1980	Murray
4,221,350 A	9/1980	Moser
4,402,478 A	9/1983	Martin
4,430,715 A	2/1984	Gentet
4,482,961 A	11/1984	Kliner et al.
4,489,123 A	12/1984	Schijve
4,500,589 A	2/1985	Schijve
4,591,213 A	5/1986	Rapoport
4,646,242 A	2/1987	Valaas
4,818,036 A	4/1989	Reinecke et al.

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See application file for complete search history.

(Continued)

FOREIGN PATENT DOCUMENTS

DE 555392 7/1932

(Continued)

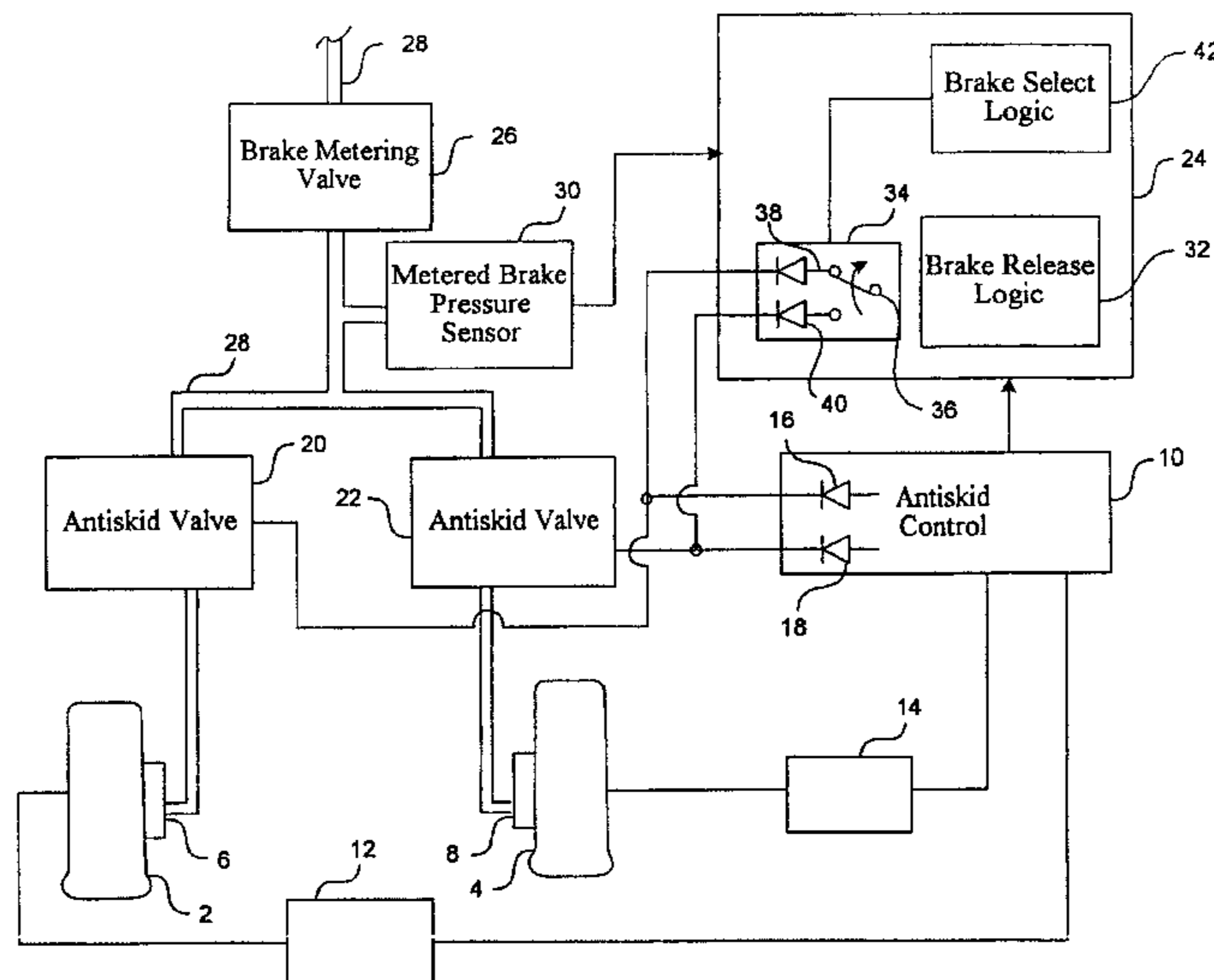
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(57) **ABSTRACT**

Methods and systems are provided for improving brake performance, including extending the life of carbon brakes on aircraft. One embodiment of the method comprises measuring the speed of the aircraft and the intensity of braking and comparing these to predetermined maximum values for each. If the values are both lower than the maximum values, one or more of the brakes are selectively disabled.

22 Claims, 2 Drawing Sheets



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U.S. PATENT DOCUMENTS

4,923,056	A	5/1990	Nedelk
4,935,291	A	6/1990	Gunnink
4,986,610	A	1/1991	Beck
4,992,323	A	2/1991	Vogelesang
5,039,566	A	8/1991	Skubic
5,160,771	A	11/1992	Lambing
5,429,326	A	7/1995	Garesche
5,547,735	A	8/1996	Roebroeks
5,615,934	A	4/1997	Abuelsamid
5,665,450	A	9/1997	Day

6,398,162	B1	6/2002	Stimson et al.
6,478,252	B1	11/2002	Stimson
6,604,708	B1	8/2003	De Vlieg
7,165,816	B2	1/2007	DeVlieg
2004/0065776	A1	4/2004	DeVlieg

FOREIGN PATENT DOCUMENTS

DE	1756129	2/1970
FR	951950	11/1949
GB	1201660	8/1970

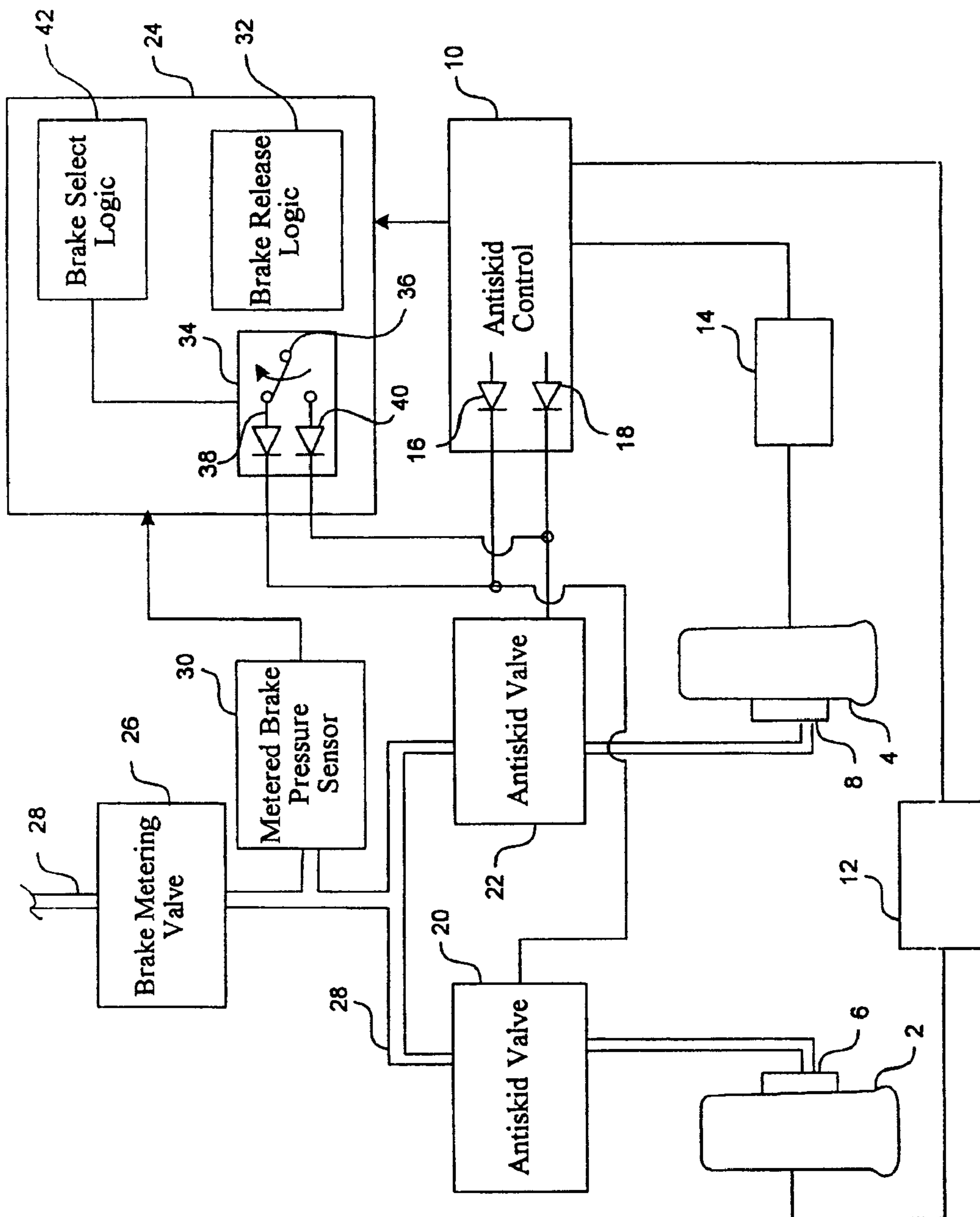


FIG. 1

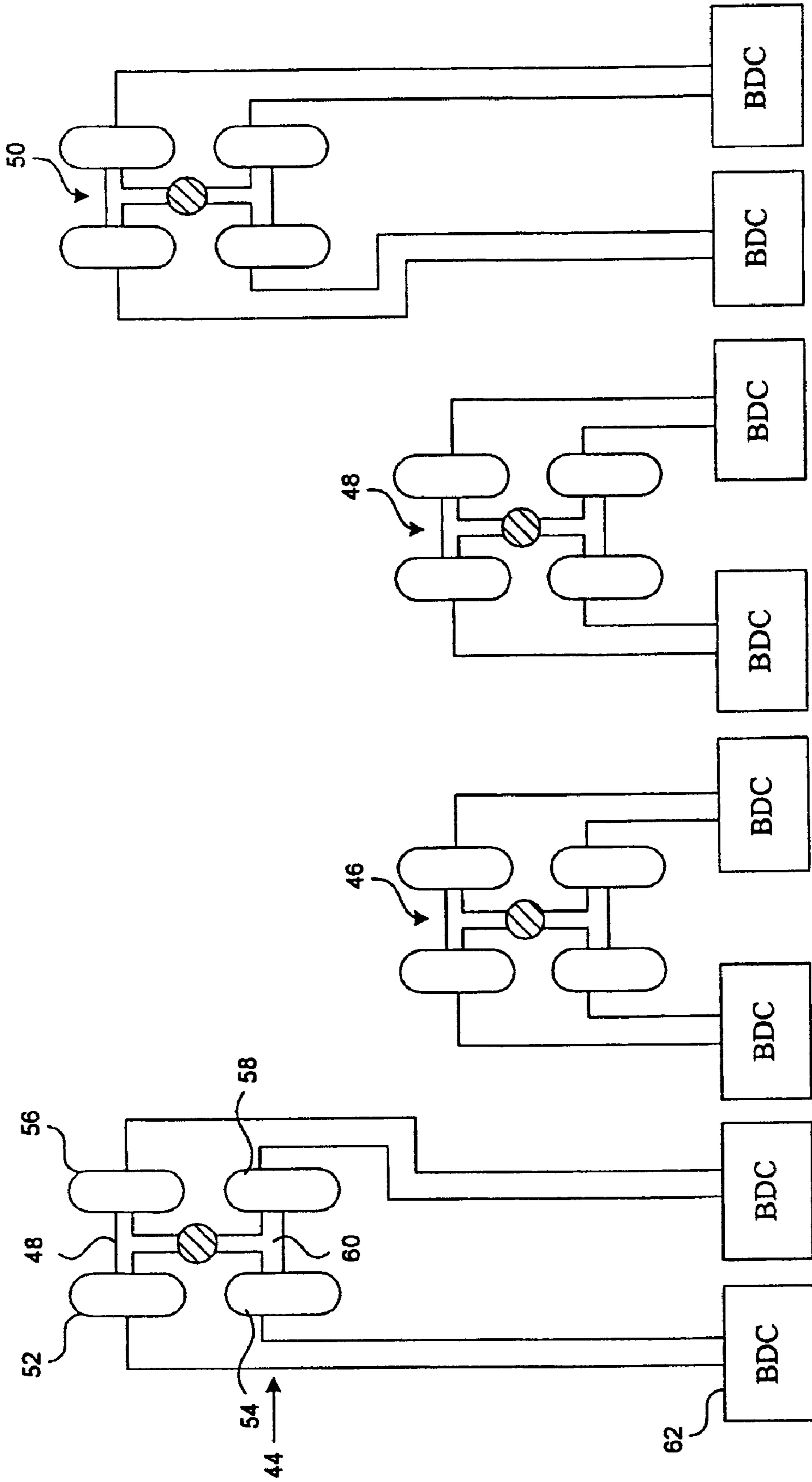


FIG. 2

CARBON BRAKE WEAR FOR AIRCRAFT

CROSS-REFERENCE TO RELATED APPLICATIONS

This application is a continuation of U.S. application Ser. No. 11/165,942, filed Jun. 24, 2005, and issued on Jan. 23, 2007 as U.S. Pat. No. 7,165,816; which is a divisional of U.S. application Ser. No. 10/617,196, filed Jul. 9, 2003, and issued on Sep. 6, 2005 as U.S. Pat. No. 6,938,857; which is a continuation of U.S. application Ser. No. 07/457,053, filed Dec. 26, 1989, and issued on Aug. 12, 2003 as U.S. Pat. No. 6,604,708.

This invention relates to a method and means for increasing the life of carbon aircraft brakes. More particularly, this invention relates to the controlled application of braking pressure to only selected braes during low speed ground travel.

BACKGROUND

Modern aircraft which are designed to carry large passenger or cargo payloads are often provided with carbon brakes on each of the wing or body mounted wheels. The nose wheel is typically not braked. Carbon brakes are preferred because of their light weight and performance characteristics and generally comprise a piston housing and parts, a torque plate and a carbon heat sink stack. This stack contains all the friction surfaces which, when compressed, cause the wheel to decrease its speed. The stack comprises a pressure plate, rotor disks, stator disks and backing plate. Carbon composite rotors are connected to the wheel through the rotor drive keys and turn with the wheel. Carbon composite stators, pressure plates and backing plate are connected to the torque tube and do not turn. Braking friction is caused when the rotors are compressed against the stators.

While carbon brakes are preferred for weight and performance reasons over steel brakes, the cost of replacing the stack divided by the number of landing cycles between replacements is much higher than for steel brakes.

Aircraft brake control systems are designed with separate pedal controls for the left and right brakes. When one of the brake pedals is depressed, all the brakes on that side of the aircraft are commanded to apply simultaneously and equally. By applying all brakes equally, the heat energy absorbed by each individual brake is minimized. For steel brakes, brake life is largely determined by the total amount of energy absorbed by each brake and is largely unaffected by the number of brake applications that accumulate that energy. Hence, brake control systems that apply all brakes simultaneously and equally provide economic operation of steel brakes and minimize exposure to overheating of any individual brake. However, direct application of this method to carbon brakes does not extend and may significantly shorten their lives. Accordingly, this invention provides a novel method and means to extend the life of carbon brakes and substantially reduce their operating cost.

BRIEF SUMMARY

In accordance with the invention carbon brake life is significantly extended by decreasing the number of brake applications during each landing cycle. More particularly, brake wear has been found to correlate significantly with the number of brake applications and to not be significantly affected by the energy absorbed during each. By far the largest number of brake applications occur during ordinary taxiing, so in preferred embodiments of this invention, only some of the

brakes are applied in response to brake applications under ordinary taxiing conditions. An alternating wheel braking pattern is established to minimize brake wear at each braked wheel and yet to promote even distribution of absorbed energy among all the brakes. This, in turn prevents overheating of any individual brake. The extended brake-wear system is activated only when aircraft ground speed and brake application pressures are typical of taxi operations. Preferably, aircraft speed and hydraulic pressure are sensed so that brakes at all wheels will be operative in critical braking situations such as landing, parking, or emergency stopping.

The invention will be better understood in terms of the Figures and detailed description which follow.

DETAILED DESCRIPTION

FIG. 1 is a simplified schematic of a subsystem for aircraft brakes which alternately disables one of two brakes in order to limit the number of brake applications and extend carbon brake life.

FIG. 2 is a schematic view of a sixteen wheel and brake landing gear configuration for a wide bodied aircraft showing a brake disable circuit which would be activated under low braking pressure and aircraft speed conditions representative of taxi braking to disable half the brakes and thereby extend brake life.

For carbon brakes, the landings to wear-out ratio is strongly dependent on the number of brake applications rather than the energy absorbed by a brake during each application. For commercial passenger aircraft, the brakes may be applied an average of twenty times per landing cycle. The brakes are generally applied during landing absorbing several million foot-pounds for heavy wide-bodied aircraft and once to stop the wheels from spinning before they are retracted after take-off. Both of these are "high speed" brake applications, and are typically at moderate hydraulic pressures less than about 1500 psi hydraulic pressure. The balance of the brake applications are "taxi snubs" for steering or low speed braking. They create hydraulic brake fluid pressures generally less than about 1500 psi and absorb about 0.5 MFP average per snub for wide-bodied aircraft. These taxi snubs account for a significant amount of brake energy temperature buildup, and for carbon brakes, most of the wear since carbon brake wear is dependent on the number of brake applications. Occasionally, "emergency" brake applications may be made at higher pressures (up to 3000 psi hydraulic fluid pressure), but such emergency braking is an insignificant wear factor.

Conventional brake wear control systems provide for applying all brakes equally, gently, and simultaneously during normal taxi braking. In accordance with a preferred embodiment of this invention, the life of carbon brakes is extended by minimizing the number of brake applications while distributing the heat energy absorbed substantially equally among all the brakes. This is accomplished by alternately applying only a selected number of brakes rather than all the brakes during each normal taxi braking operation.

A simplified example of a preferred embodiment of this invention is shown schematically in FIG. 1. A left wheel 2 and right wheel 4 are on the same side of an airplane and are actuated by the one of the two brake pedals in the cockpit. Wheel 2 has a carbon brake 6 and right wheel 4 a carbon brake 8. In this embodiment, the antiskid control system 10 is integral to the brake disable system. Left and right wheel speed sensors 12 and 14, electronically measure wheel speeds and input the signals generated to the antiskid control circuit 10. Signals from antiskid control circuit 10 are outputted through diodes 16 and 18 to left and right hydraulic antiskid valves 20

and 22. The signals from wheel speed sensors 12 and 14 are integrated by antiskid control circuit 10 and outputted to brake disable control circuit 24.

Brake metering valve 26 which is responsive to a call for braking from the cockpit is located in brake hydraulic line 28. The static line pressure is low, pressure during taxi snubs is higher, and pressure during parking and emergency braking is relatively higher still. This pressure is measured at metered brake pressure sensor 30. The signal from sensor 30 is inputted to brake disable control circuit 24.

The system works in accordance with the invention as follows. The speeds of wheels 2 and 4 are sensed through sensors 12 and 14 and processed in antiskid control circuit 10 to determine aircraft speed. That aircraft speed signal is inputted to brake disable circuit 24. The desired intensity of braking action is sensed by the metered brake pressure sensor 30 and is also inputted to brake disable circuit 24. Inside brake disable control circuit 24, the metered pressure signal is compared against a first predetermined value, 100 psi for example, to detect when a brake application has been commanded. At the moment at which a brake application is detected, a comparison is made between the aircraft speed signal and a predetermined value for aircraft speed in brake release logic circuit 32. If the speed is higher than the predetermined value, 40 mph, for example, then brake disablement is not enabled. Subsequently, comparison is continuously made inside brake release logic 32 between the metered pressure signal value and a second predetermined value. If the pressure is greater than the second predetermined value, greater than 1500 psi, for example, then the brake disable control circuit 24 does not disable any brakes. That is, if heavy braking intensity is called for, all the brakes are applied. If and only if aircraft speed at the time of brake application and metered brake pressure are lower than their predetermined maximum values will brake release logic circuit 42 be activated.

As indicated by bipolar knife switch 36, only one of the two antiskid valves 20 and 22 will be commanded to release its respective brake through left diode 38 or right diode 40 when brake release logic 32 triggers. Brake select logic circuit 42 remembers which brake was last disabled and switches switch 36 when a new brake application has been detected by brake disable circuit 24.

Brake disable logic 24 responds to both the metered pressure signal and the aircraft speed signal at the time of brake application. Thereafter, logic circuit 24 responds only to the metered pressure signal from sensor 30 until the metered brake pressure returns to the no-braking system pressure. This ensures that following a high speed brake application, such as a landing, the brake release command will not be produced, and half the brakes will not be released, as the aircraft decelerates through the brake disable speed threshold. The disable signal would then only be produced at low speed after the brakes were released, then reapplied.

If an emergency stop, i.e., high metered pressure is sensed by brake disable circuit 24, then brake release logic 32 removes the brake release command so that both brakes 6 and 8 are applied, thus insuring full aircraft braking capability when it is needed. Similarly, if a higher speed stop, such as a landing stop or rejected take off, is sensed by brake disable circuit 24 from the aircraft speed signal, then the brake release logic 32 removes the brake release command so that both brakes 6 and 8 may share the braking energy, preventing overheating of an individual brake or brakes.

While the desired braking intensity has been described in terms of metered braking pressure, other input to the brake disable circuit providing like information would be equally

useful. For example, the acceleration and throw of the brake pedal in the cockpit could be monitored or the rate of brake temperature increase. Similarly, input other than aircraft speed such as wheel speed or aircraft ground speed measured independently of the wheel speed could be inputted to the brake disable circuit. Such alternatives will be apparent to those skilled in the art.

The invention has been described specifically in FIG. 1 in terms of a brake pair on one side of an aircraft. However, systems in accordance with this invention for aircraft with other numbers and arrangements of carbon braked wheels could be readily adapted by persons skilled in the art. For example, FIG. 2 shows the wheel configuration for a wide-bodied Boeing 747-400™ series aircraft equipped with a carbon brake on each main gear wheel. The nose wheel which is not braked is not shown.

Referring to FIG. 2, there are four four-wheel trucks located under the left wing 44, left body 46, right body 48 and right wing 50 of an aircraft. Using truck 44 as an example, wheels 52 and 54 on one side, and 56 and 58 on the other side of a four-wheel axle frame 60 each provide input to a brake disable circuit 62 like that described in FIG. 1. A metered brake pressure signal would also be provided to each like brake disable circuit. Thus, when both the aircraft speed at time of brake application and metered brake pressure are below target values, half of the sixteen brakes would be disabled. For example, brakes on wheels 52 and 54 on the left side of the truck 60 would be alternately disabled during successive brake applications as would the brakes on wheels 56 and 58.

Since Carbon brake wear is a function of the number of applications, and since the vast majority of brake applications occur during taxiing, the life of carbon brakes is significantly improved by practicing this invention. For example, if half the brakes are applied during each taxi brake application, brake wear life could nearly double. The life of carbon brakes might be proportionately extended even further by disabling even more than half the brakes during each braking cycle. System logic insures maximum braking capability during emergency braking, i.e., high pressure, conditions. Overheating of individual brakes is prevented because system logic alternates between brakes to share the braking energy among all the brakes.

Other system refinements such as redundant metered pressure sensors could be added to improve failure mode performance. Also, means could be provided to smooth brake pedal control responsiveness in the cockpit between partial brake and full brake transitions. That is, the back pressure on the brake pedal could be adjusted so that equal pedal depression results in equal braking responsiveness irrespective of how many brakes are being disabled at a given time. Brake temperature could also be considered in the brake disabling algorithm to prevent disablement if some brakes are too hot from previous brake applications.

While the invention has been described in terms of specific embodiments thereof, other forms may be readily adapted by one skilled in the art. Accordingly, the scope of the invention is to be limited only in accordance with the following claims.

The invention in which an exclusive property or privilege is claimed are defined as follows:

1. A method of controlling brakes of a multiple brake vehicle, comprising:
 - receiving a signal corresponding to a speed of the vehicle when braking;
 - receiving a signal corresponding to a desired braking intensity;

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comparing the speed of the vehicle with a preset value for the vehicle speed;

comparing the desired braking intensity with a preset range for braking intensity at least twice during braking; and

if, at least approximately the time a brake application is detected, the speed is below the preset value for the vehicle speed and the desired braking intensity is outside the preset range for the braking intensity, directing a signal to disable at least one brake during braking and thereafter directing a signal to selectively disable a different brake during a succeeding brake application.

2. The method of claim 1, further comprising detecting the speed of the vehicle when braking, and transmitting the signal corresponding to the speed of the vehicle when braking.

3. The method of claim 1, further comprising detecting the desired braking intensity and transmitting the signal corresponding to the desired braking intensity.

4. The method of claim 1, wherein comparing the desired braking intensity at least twice during braking includes comparing the desired braking intensity continuously throughout braking.

5. The method of claim 1, further comprising selectively disabling at least one of the brakes.

6. The method of claim 1 wherein directing a signal to selectively disable at least one of the brakes includes directing a signal to selectively disable half the brakes of the vehicle.

7. The method of claim 1 wherein directing a signal to selectively disable at least one of the brakes includes directing a signal to selectively disable more than half the brakes of the vehicle.

8. The method of claim 1, further comprising:
receiving a signal corresponding to a brake temperature;
and
if the brake temperature is above a predetermined value, preventing disabling of the at least one brake.

9. The method of claim 1 wherein directing a signal to disable at least one brake includes directing the signal when the desired braking intensity is below a preset value for the braking intensity.

10. The method of claim 1, further comprising directing a signal to re-enable the at least one disabled brake upon receiving an indication that the desired braking intensity is within the present range for the braking intensity.

11. The method of claim 1 wherein receiving a signal corresponding to a speed of the vehicle includes receiving a signal corresponding to the speed of an aircraft.

12. The method of claim 1 wherein comparing the speed of the vehicle with a preset value includes comparing the speed of the vehicle only at least approximately the moment at which a brake application is detected.

13. An apparatus, comprising:

a computer-based system for controlling brakes of a multiple brake vehicle, the system including:

a first receiver portion configured to receive a signal corresponding to a speed of a vehicle when the vehicle is braking;

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a second receiver portion configured to receive a signal corresponding to a desired braking intensity;

a first comparer portion configured to compare the signal corresponding to the speed of the vehicle with a predetermined value;

a second comparer portion configured to compare the signal corresponding to the desired braking intensity with a predetermined range for braking intensity at least twice during braking; and

a disabling portion configured to direct disablement of at least one of the brakes if, at least approximately the time a brake application is detected, the speed is below the predetermined value for the vehicle speed and the desired braking intensity is outside the predetermined range for the braking intensity, the braking portion further being configured to direct disablement of another of the brakes during a succeeding brake application.

14. The apparatus of claim 13 wherein the disabling portion is configured to direct disablement of at least one of the brakes when the desired braking intensity is below a predetermined value for the braking intensity.

15. The apparatus of claim 13, further comprising a speed detector positioned to detect the speed of the vehicle when braking and transmit the signal corresponding to the speed of the vehicle when braking.

16. The apparatus of claim 13, further comprising a braking intensity detector positioned to detect the desired braking intensity and transmit the signal corresponding to the desired braking intensity.

17. The apparatus of claim 13 wherein the second comparer portion is configured to compare the signal corresponding to the desired braking intensity continuously throughout braking.

18. The apparatus of claim 13 wherein the disabling portion is configured to direct disablement of half the brakes of the vehicle.

19. The apparatus of claim 13 wherein the disabling portion is configured to direct disablement of more than half the brakes of the vehicle.

20. The apparatus of claim 13, further comprising the vehicle.

21. The apparatus of claim 13, further comprising the vehicle, and wherein the vehicle includes an aircraft having a fuselage, wings depending from the fuselage, and landing gear, with the system for controlling brakes being operatively coupled to the landing gear.

22. The apparatus of claim 13 wherein the first comparer portion is configured to compare the signal corresponding to the speed of the vehicle with a predetermined value only at least approximately the moment at which a brake application is detected.

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