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(54) **ROTOR BLADE CONTAINMENT ASSEMBLY FOR A GAS TURBINE ENGINE**

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F01D 21/00 (2006.01)

(52) **U.S. Cl.** **415/9; 415/214.1**

(58) **Field of Classification Search** **415/9, 415/173.4, 214.1**

See application file for complete search history.

(56) **References Cited**

U.S. PATENT DOCUMENTS

4,293,053 A 10/1981 Shuttleworth et al.
5,201,846 A 4/1993 Sweeney

5,267,828 A * 12/1993 Lenhart et al. 415/9
5,431,532 A * 7/1995 Humke et al. 415/9
5,885,056 A * 3/1999 Goodwin 415/9
6,468,026 B1 10/2002 Bonnoitt et al.
2005/0089391 A1* 4/2005 Stretton 415/9

FOREIGN PATENT DOCUMENTS

EP 1 013 894 A2 6/2000
EP 1 087 104 A2 3/2001
EP 1 245 794 A2 10/2002

* cited by examiner

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(57) **ABSTRACT**

A gas turbine engine rotor blade containment assembly (38) comprising a generally cylindrical, or frustoconical, casing (40). The casing (40) having at least one member (54) and the member (54) has a first portion (53) extending radially from the casing (40) and a second portion (55) extending axially from the first portion (53) of the member (54). At least one cassette (80) being provided and one or more liner panels (74) being arranged in each cassette (80). A first end (84) of each cassette (80) being locatable on the second portion (55) of the member (54) and the first end (84) of each cassette (80) being shaped to correspond to the shape of the first and second portions (53,55) of the at least one member (54) to protect the first and second portions (53,55) of the at least one member (54). The second end (86) of each cassette (80) having at least one axially extending member (96) removably securable to the casing (40).

15 Claims, 3 Drawing Sheets

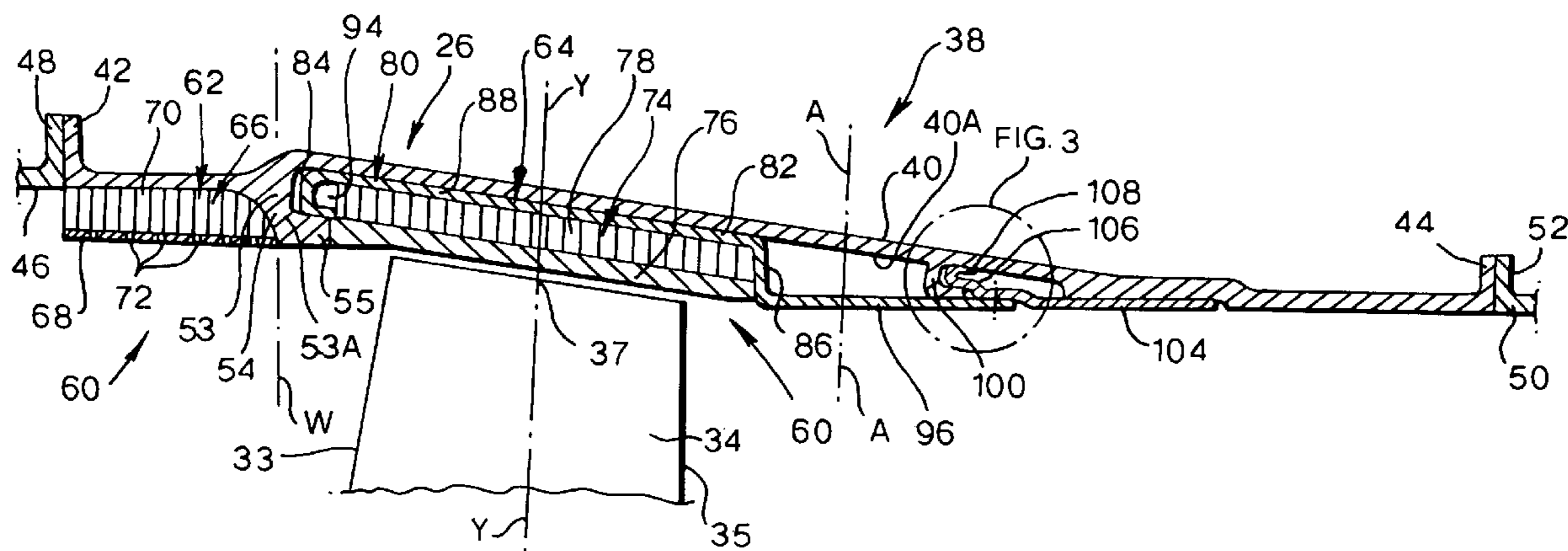


Fig. 1.

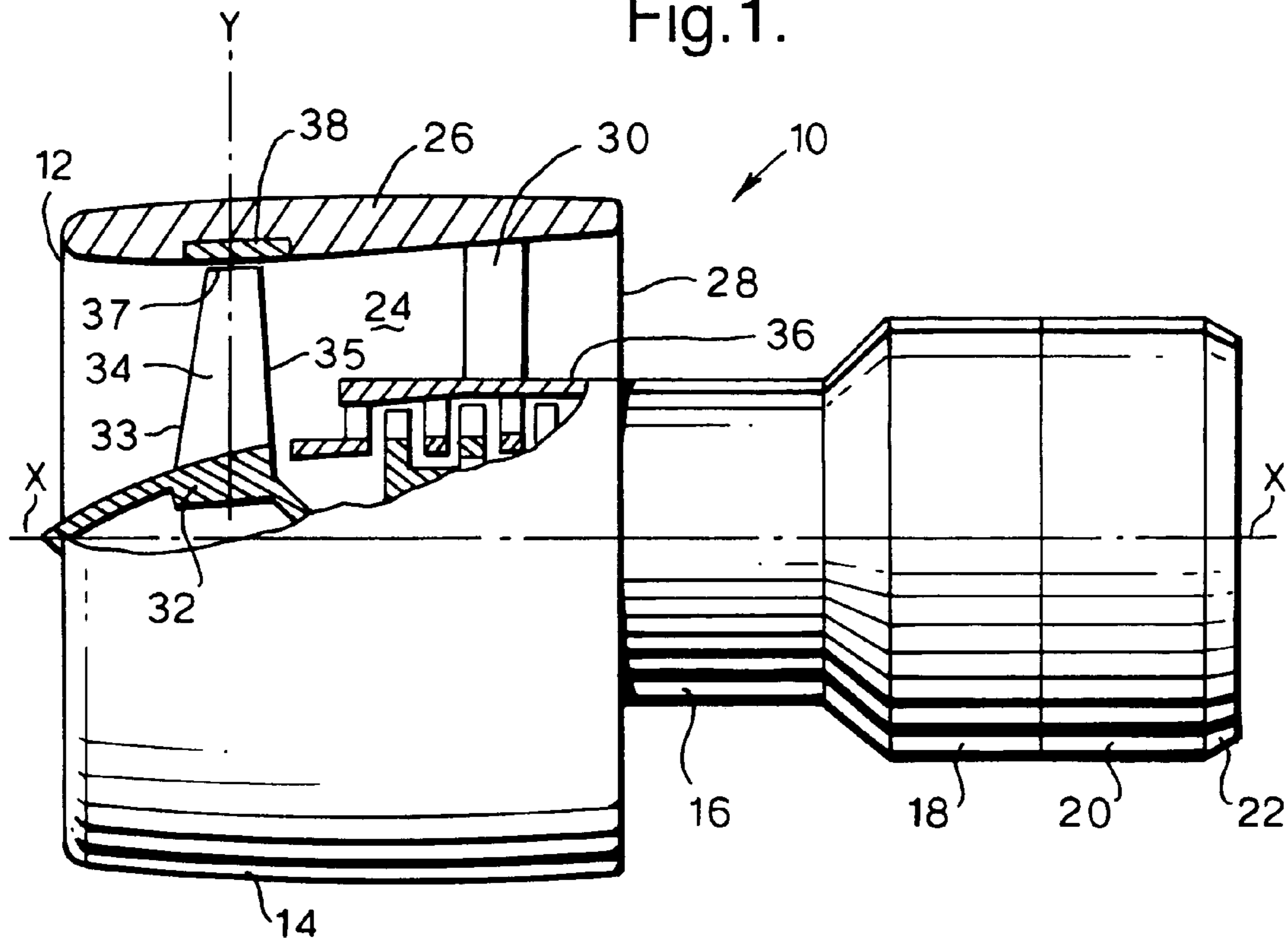


Fig. 5.

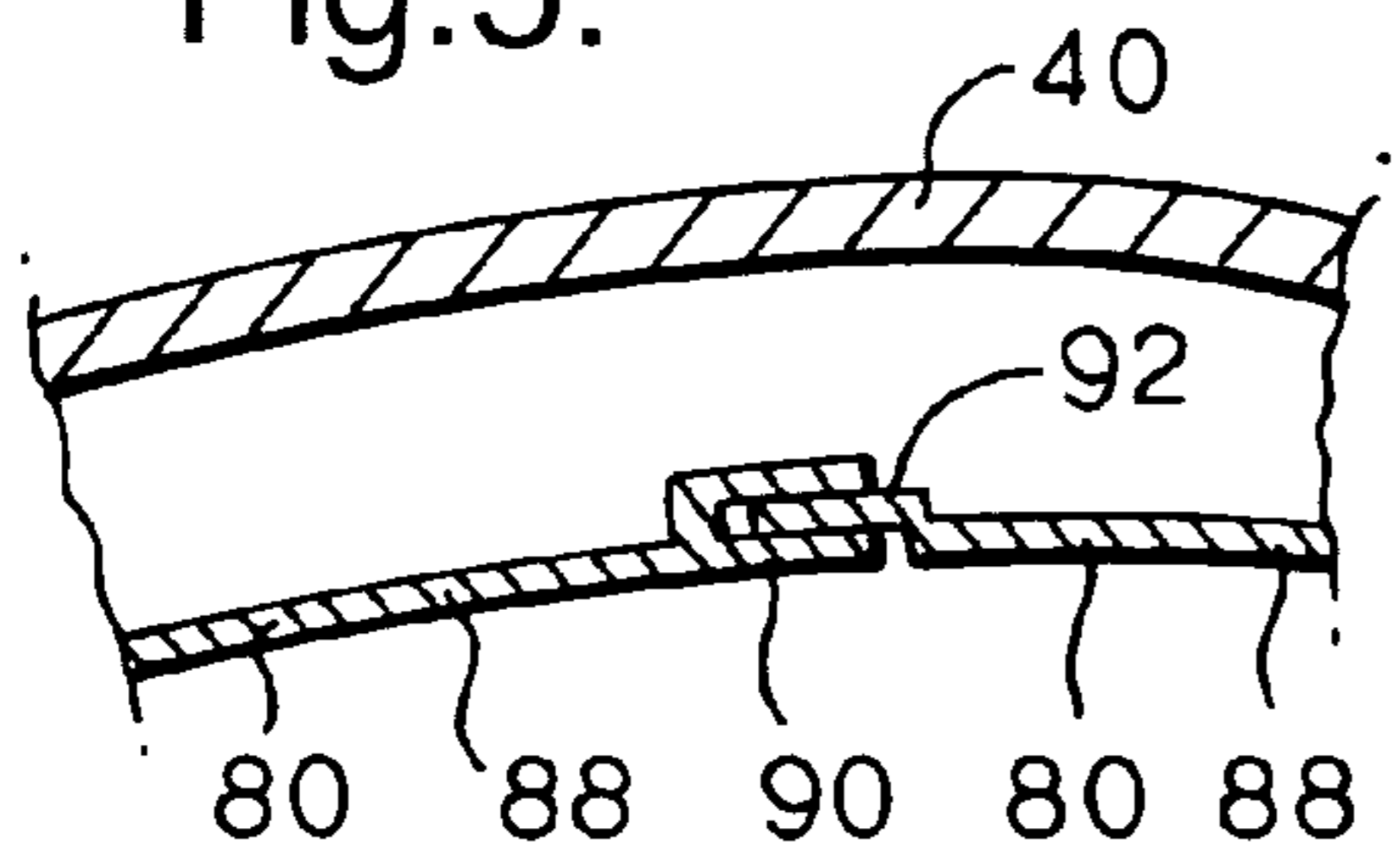


Fig. 6.

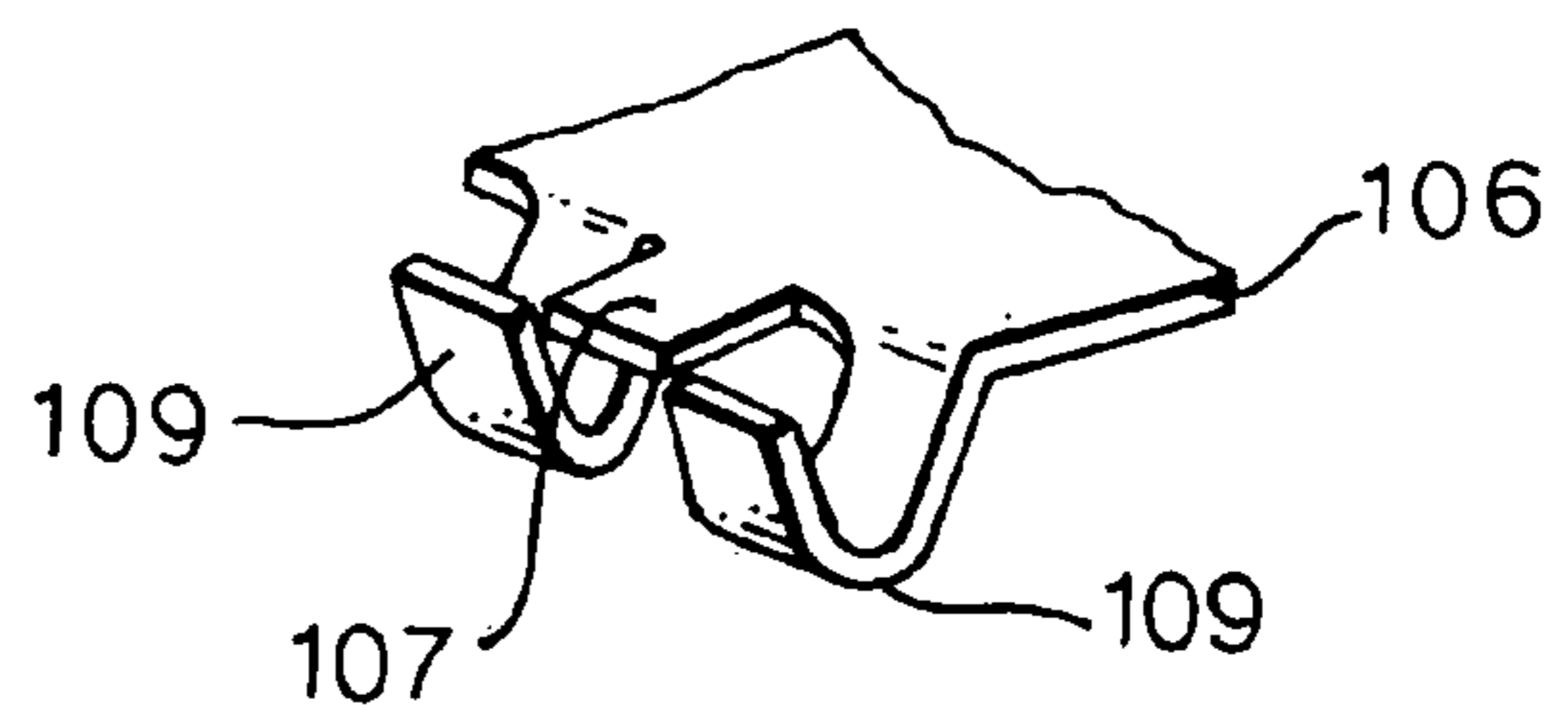


Fig.7.

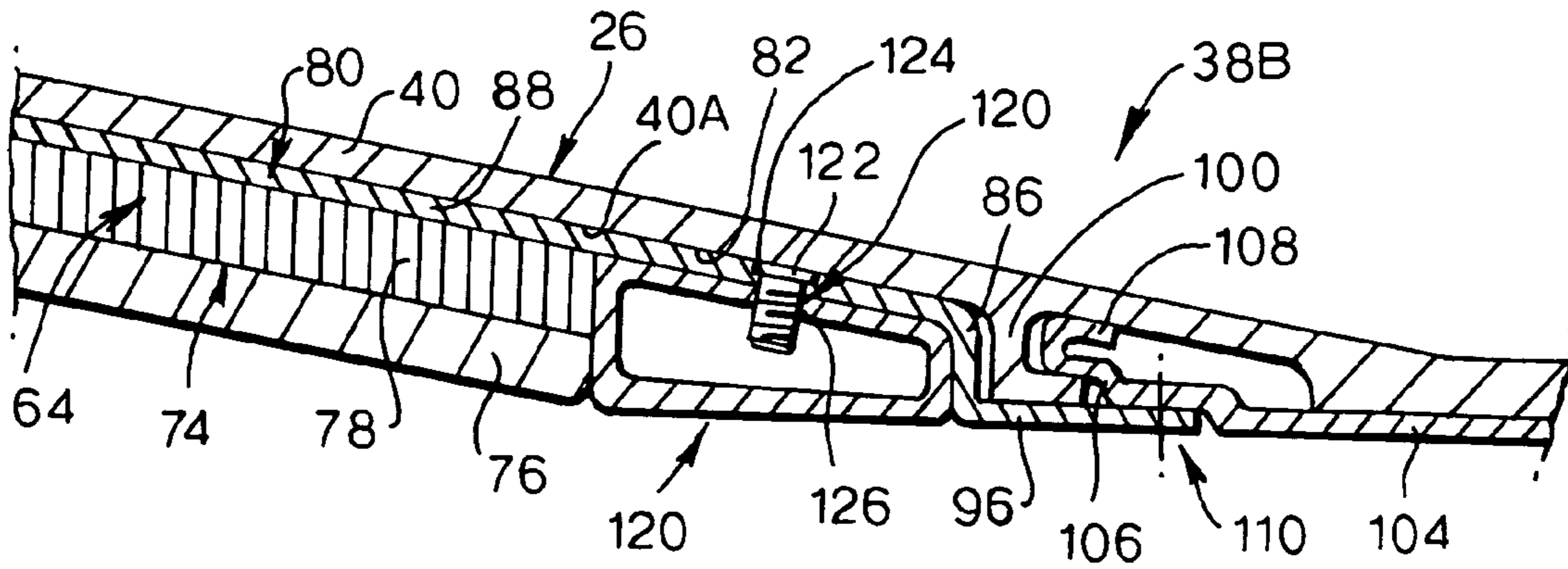


Fig.8.

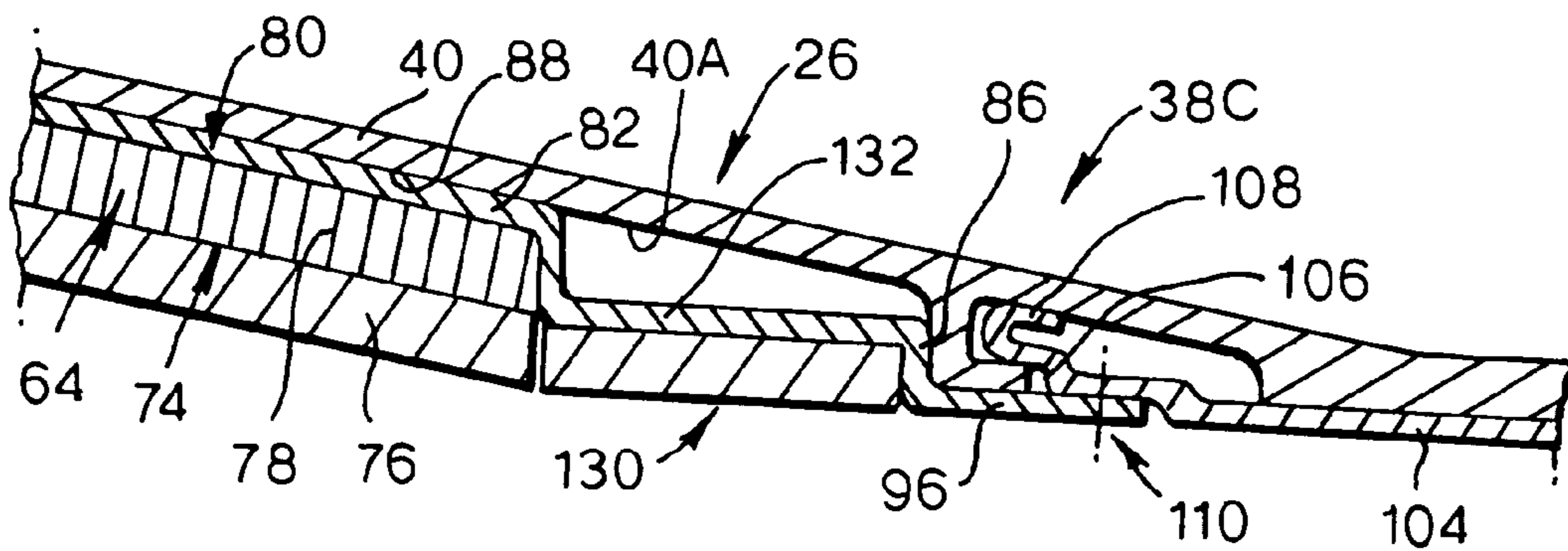
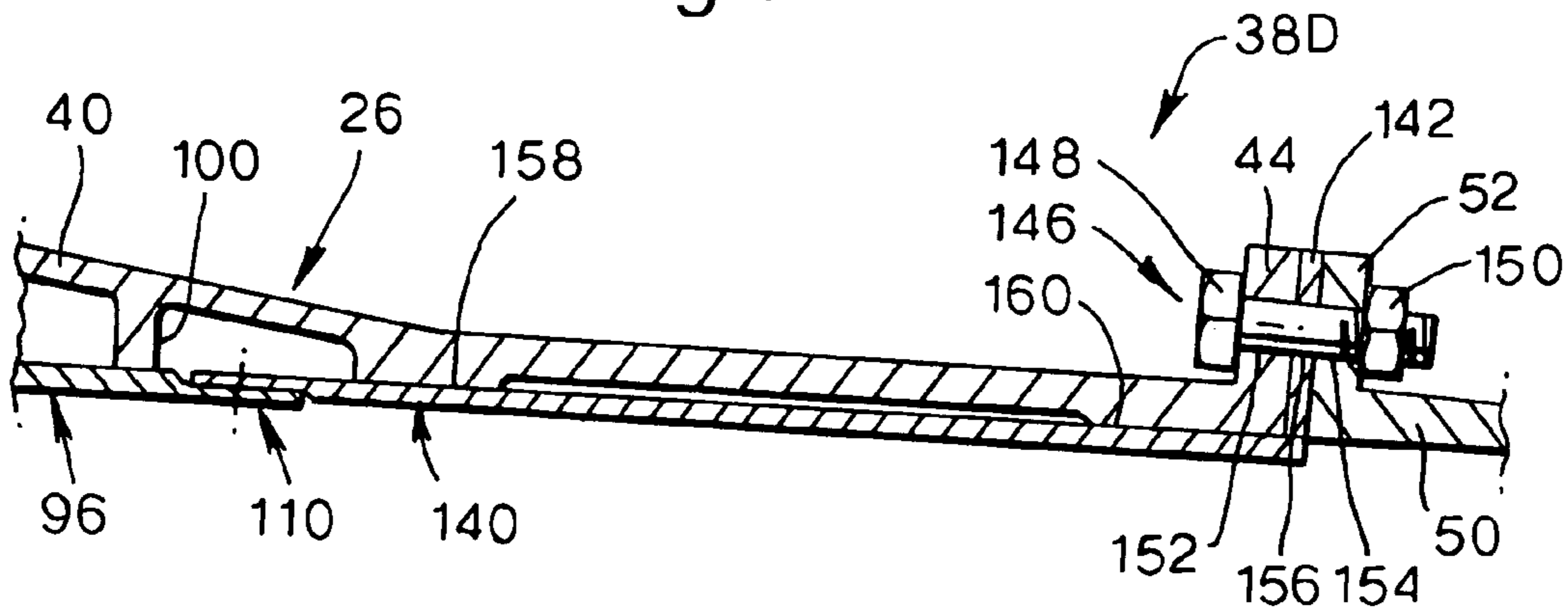


Fig.9.



ROTOR BLADE CONTAINMENT ASSEMBLY FOR A GAS TURBINE ENGINE

The present invention relates to a rotor blade containment assembly for a gas turbine engine, particularly to a fan blade containment assembly or a turbine blade containment assembly for a gas turbine engine.

Turbofan gas turbine engines for powering aircraft conventionally comprise a core engine, which drives a fan. The fan comprises a number of circumferentially spaced radially outwardly extending fan blades mounted on a fan rotor enclosed by a generally cylindrical, or frustoconical, fan casing. The core engine comprises one or more turbines, each one of which comprises a number of circumferentially spaced radially outwardly extending turbine blades enclosed by a cylindrical, or frustoconical, casing.

There is a remote possibility with such engines that part, or all, of a fan blade, or a turbine blade, could become detached from the remainder of the fan or turbine. In the case of a fan blade becoming detached this may occur as the result of, for example, the turbofan gas turbine engine ingesting a bird or other foreign object.

The use of containment rings for turbofan gas turbine engine casings is well known. It is known to provide generally cylindrical, or frustoconical, relatively thick metallic containment rings. It is also known to provide generally cylindrical, or frustoconical, locally thickened, isogrid, metallic containment rings. Furthermore, it is known to provide strong fibrous material wound around relatively thin metallic casings or around the above-mentioned containment casings. In the event that a blade becomes detached it passes through the casing and is contained by the fibrous material.

The fan casing is provided with a hook axially upstream of the tips of the fan blades to prevent upstream movement of a detached fan blade.

A fan blade track liner is provided within the fan casing around and adjacent the tips of the fan blades.

Conventionally the fan blade track liner is secured within the fan casing by adhesive bonding.

Our UK patent application 0324549.5 filed 22 October describes an arrangement where one or more fan blade track liner panels are arranged in a cassette and the cassette has an axially extending member locatable on the hook. The cassette comprises for example a composite material. The cassette merely holds the fan blade track liner panels on and adjacent the fan casing. In the event that a fan blade becomes detached it passes through the cassette and is contained by the casing.

Accordingly the present invention seeks to provide a novel gas turbine engine rotor blade containment assembly.

Accordingly the present invention provides a gas turbine engine rotor blade containment assembly comprising a generally cylindrical, or frustoconical, casing, the casing having at least one member, the at least one member having a first portion extending radially from the casing and a second portion extending axially from the first portion of the at least one member, at least one cassette, one or more liner panels being arranged in the at least one cassette, a first end of the at least one cassette being locatable on the second portion of the at least one member, the first end of the at least one cassette being shaped to correspond to the shape of the first and second portions of the at least one member to protect the first and second portions of the at least one member, the second end of the at least one cassette having at least one axially extending member removably securable to the casing.

Preferably there is a plurality of cassettes.

Preferably one or more liner panels being arranged in each cassette, a first end of each cassette being locatable on the

second portion of the member, the first end of each cassette being shaped to correspond to the shape of the first and second portions of the at least one member to protect the first and second portions of the at least one member, the second end of each cassette having at least one axially extending member removably securable to the casing.

Preferably the cassettes are arranged circumferentially around and within the casing.

Preferably the cassettes have axially extending edges, the axially extending edges of adjacent cassettes overlap.

The axially extending edge of one of the adjacent cassettes may have a tongue and the axially extending edge of the other of the adjacent cassettes may have a groove.

Preferably the first end of the at least one cassette is U-shape or C-shape in cross-section.

Preferably the at least one member is annular.

Preferably the casing comprises a metal, more preferably the casing comprises titanium, aluminium, an aluminium alloy, a titanium alloy, magnesium or magnesium alloy.

Preferably the at least one cassette comprises a metal, more preferably the at least one cassette comprises steel, titanium or a titanium alloy.

Preferably the at least one axially extending member of the at least one cassette being removably securable to the casing by at least one retention member, the at least one retention member being removably securable to the casing.

Preferably the casing is a fan casing and the rotor blades are fan blades. The casing may be a compressor casing and the rotor blade may be compressor blades. The casing may be a turbine casing and the rotor blades may be turbine blades.

The present invention will be more fully described by way of example with reference to the accompanying drawings in which:

FIG. 1 is a partially cut away view of a gas turbine engine having a fan blade containment assembly according to the present invention.

FIG. 2 is an enlarged cross-sectional view of the fan blade containment assembly shown in FIG. 1.

FIG. 3 shows a further enlarged cross-sectional view of part of the fan blade containment assembly shown in FIG. 2.

FIG. 4 is a cross-section along line A-A through the fan blade containment assembly shown in FIG. 2.

FIG. 5 is an alternative cross-section along line A-A through the fan blade containment assembly shown in FIG. 2.

FIG. 6 is a further enlarged perspective view of an alternative part of the fan blade containment assembly shown in FIG. 2.

FIG. 7 is an alternative enlarged cross-sectional view of the fan blade containment assembly shown in FIG. 1.

FIG. 8 is another alternative enlarged cross-sectional view of the fan blade containment assembly shown in FIG. 1.

FIG. 9 is another alternative enlarged cross-sectional view of the fan blade containment assembly shown in FIG. 1.

A turbofan gas turbine engine **10**, as shown in FIG. 1, comprises in axial flow series and intake **12**, a fan section **14**, a compressor section **16**, a combustor section **18**, a turbine section **20** and an exhaust **22**. The turbine section **22** comprises one or more turbines (not shown) arranged to drive one or more compressors (not shown) in the compressor section **16** via shafts (not shown). The turbine section **20** also comprises one or more turbines (not shown) arranged to drive the fan section **14** via a shaft. The fan section **14** comprises a fan duct **24** defined partially by a fan casing **26**. The fan duct **24** has an outlet **28** at its axially downstream end. The fan casing **26** is secured to a core engine casing **36** by a plurality of circumferentially spaced radially extending fan outlet guide vanes **30**. The fan casing **26** surrounds a fan rotor **32**, which

carries a plurality of circumferentially spaced radially outwardly extending fan blades **34**. The fan rotor **32** and fan blades **34** rotate about an axis X of the gas turbine engine **10**, substantially in a plane Y perpendicular to the axis X. The fan casing **26** also comprises a fan blade containment assembly **38**, which is arranged substantially in the plane Y of the fan blades **34**. The fan blades **34** have a leading edge **33**, a trailing edge **35** and a tip **37**.

The fan casing **26** and fan blade containment assembly **38** is shown more clearly in FIGS. **2**, **3**, **4** and **5**. The fan blade containment assembly **38** comprises a metal, cylindrical, or frustoconical, casing **40**. The metal casing **40** comprises an upstream flange **42** by which the fan blade containment assembly **38** is connected to a flange **48** on an intake assembly **46** of the fan casing **26**. The metal casing **40** also comprises a downstream flange **44** by which the fan blade containment assembly **38** is connected to a flange **52** on a rear portion **50** of the fan casing **26**. The metal casing **40** comprises an annular member, or hook, **54** positioned axially upstream of the leading edge **33** of the tip **37** of the fan blade **34**. The annular member **54** is L-shaped in cross-section in planes containing the axis X of the gas turbine engine **10** and also of the metal casing **40**, and the annular member **54** comprises a first portion **53** which extends in a radially inwardly direction from the metal casing **40** and a second portion **55** which extends in an axially downstream direction from the radially inner end **53A** of the first portion **53** of the annular member **54** towards the tip **37** of the fan blade **34**. The annular member **54** is substantially in a plane W, perpendicular to the axis X of the gas turbine engine **10**, upstream of the most upstream point of the leading edge **33** of the fan blade **34**.

A liner **60** is provided radially within the metal casing **40** and there are two types of liner **60**, an acoustic liner **62** to reduce noise and a fan blade track liner **64** to form an abradable seal.

The acoustic liner **62** generally comprises a plurality of acoustic panels **66**, which are arranged circumferentially and axially along the inner surface **40A** of the metal casing **40**. Each acoustic panel **66** comprises a perforated skin **68** and a structure **70** to form an acoustic treatment structure. The perforated skin **68** has a plurality of perforations **72**. The perforated skin **68** defines the flow path through the fan duct **24**.

The perforated skin **68** of each acoustic panel **66** comprises aluminium, titanium or composite material, for example fibre reinforced plastic e.g. glass fibre reinforced epoxy resin. The structure **70** comprises a honeycomb structure, for example an aluminium honeycomb, a titanium honeycomb, a composite material honeycomb, a resin impregnated paper honeycomb or other suitable honeycomb. The composite material honeycomb may comprise fibre reinforced plastic e.g. glass fibre reinforced epoxy resin.

The fan blade track liner **64** generally comprises a plurality of fan blade track panels **74** which are arranged circumferentially and axially along the inner surface **40A** of the metal casing **40**. Each fan blade track panel **74** comprises a skin **76** and a structure **78** to form an abradable structure. The skin **76** defines the flow path through the fan duct **24**.

The skin **76** of each fan blade track panel **64** comprises for example a Nomex (RTM) aramid fibre paper honeycomb structure filled with abradable filler. The abradable filler comprises for example hollow microspheres in an epoxy paste. However, other suitable honeycombs and abradable fillers may be used. The structure **78** comprises a honeycomb structure, for example an aluminium honeycomb, a titanium honeycomb, a composite material honeycomb, a resin impregnated paper honeycomb or other suitable honeycomb. The

composite material honeycomb may comprise fibre reinforced plastic e.g. glass fibre reinforced epoxy resin.

The acoustic panels **66** are secured to the metal casing **40** by suitable means known in the art for example adhesive bonding, fasteners etc.

The fan blade track panels **74** are secured to a plurality of cassettes **80** by adhesive bonding, for example each cassette **80** has one or more fan blade track panels **74**. Each cassette **80** comprises an open ended and open sided box structure **82** and in particular the box structure comprises a base portion **88**, an upstream end **84** and a downstream end **86**. The base portion **88** has edges **90** and **92** extending axially between the upstream end **84** and the downstream end **86** of the cassette **80**. The cassettes **80** are arranged circumferentially within the metal casing **40** around the fan blades **34** substantially in the plane Y. The base portion **88** is shaped to conform to the shape of inner surface **40A** of the metal casing **40**.

Each cassette **80** is formed from metal, for example steel, and is preferably formed from steel with high strength characteristics. But other suitable metals, for example titanium or a titanium alloy may be used. Alternatively each cassette **80** may comprise a ceramic material.

The upstream end **84** of each cassette **80** is arranged to extend radially inwardly from the base portion **88** and then in an axially downstream direction such that it is L-shaped, hook-shaped, in cross-section in planes containing the axis X of the gas turbine engine **10** and also of the metal casing **40**, to form a continuous channel **94** with the base portion **88**. The channel **94** is open at its downstream end and faces towards the downstream end **86** of the cassette **80**. In addition each cassette **80** is arranged such that the L-shaped, or hook-shaped, upstream end **84** is dimensioned and shaped to match the first portion **53** and second portion **55** of the annular L-shaped member **54**. The radially inner end of the upstream end **84** of each cassette **80** is radially outward of the axially extending second portion **55** of the annular member **54** and thus secures the upstream end **84** of each cassette **80** to the metal casing **40**. The upstream end **84** of each cassette **80** is locatable on the second portion **55** of the annular L-shaped member **54**. During a fan blade off event, the upstream end **84** of each cassette **80** protects the annular L-shaped member **54** when the tip of a detached fan blade **34** moves in an upstream direction axially along the metal casing **40**, and protects the annular L-shaped member **54** from being cut off the metal casing **40** by the detached fan blade **34**.

The adjacent edges **90** and **92** of the cassettes **80** are arranged to overlap, either as a lap, as shown in FIG. **4**, or as a tongue and groove, as shown in FIG. **5**. The overlapping of the adjacent edges **90** and **92** of the base portions **88** of the cassettes **80** is arranged such that during a fan blade off event, the tip **37** of a rotating fan blade **34** rides along the overlap instead of engaging the ends **90** and **92**.

A compliant layer, for example polysulphide, may be provided between the cassettes **80** and the metal casing **40** to reduce fretting between the cassettes **80** and the metal casing **40**.

The metal casing **40** has a second annular member, or hook, **100**, as shown more clearly in FIG. **3**. The second annular member **100** is L-shaped in cross-section in planes containing the axis X of the gas turbine engine **10** and also of the metal casing **40** and comprises a first portion **101** which extends in a radially inwardly direction from the metal casing **40** and a second portion **103** which extends in an axially downstream direction from the radially inner end **101A** of the first portion **101** of the second annular member **100**. A number of part annular retention members **104**, for example two half annular members, are provided to retain the downstream ends **86** of

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the cassettes **80** on the metal casing **40**. Preferably each retention member **104** is formed from metal, for example steel, and is preferably formed from steel with high strength characteristics. But other suitable metals, for example titanium or a titanium alloy may be used. The upstream end **106** of each retention member **104** has a hook **108** which is arranged to locate radially outwardly of the second portion **103** of the second annular member **100** on the metal casing **40**. The hook **108** is preferably U-shaped, or C-shaped, in cross-section in planes containing the axis X of the gas turbine engine **10** and also of the metal casing **40**. The axially extending edges of the adjacent retention members **104** are also arranged to overlap, as a lap, similarly to FIG. 4 and the edges of the adjacent retention members **104** are secured together, by fasteners, e.g. bolts, or screws, and nuts or rivets.

The downstream end **86** of each cassette **80** has an axially extending member **96** to secure the cassettes **80** to the metal casing **40**. The axially extending member **96** at the downstream end **86** of each cassette **80** is secured to one of the retention members **104** by fastening means **110**, for example one or more radially extending screws **112** and anchor nuts **114**, extending through countersunk apertures **116** in the axially extending member **96** of the cassettes **80** and apertures **118** in the retention members **104**. In addition the radially outer surface of the axially extending member **96** of each cassette **80** abuts the second portion **103** of the second annular member **100**.

The fan blade track panels **74** are installed by firstly adhesively bonding the fan blade track panels **74** to the cassettes **80**. The retention members **104** are installed by moving them in an axially upstream direction so that the upstream ends **106** of the retention members **104** locate on the radially outer surface of the second portion **103** of the second annular member **100**. The edges of the retention members **104** are arranged to overlap and then the edges are secured together, for example by fasteners, e.g. bolts, or screws, and nuts or rivets. The cassettes **80** are installed by moving them in an axially upstream direction so that the upstream ends **84** of the cassettes **80** locate on the radially outer surface of the second portion **55** of the annular member **54**. The cassettes **80** are then moved radially outwardly to abut the radially inner surface **40A** of the metal casing **40** and so that the axially extending members **96** abut the second portions **103** of the annular members **100** and the retention members **104**. Then the screws, or bolts, **112** are inserted through the apertures **116** and **118** in the axially extending members **96** and the retention members **104** and threaded into the anchor nuts **114**. The apertures **118** are preferably slots, e.g. axially elongate in an axial direction, to protect the fasteners from shearing in the event of fan blade **34** becoming detached.

The fan track liner panels **74** are removed by unfastening the screws, or bolts, **112** from the anchor nuts **114** and removing the screws **112**. The cassettes **80** are removed by moving them in an axially downstream direction so that the upstream ends **84** no longer locate on the radially outer surface of the second portion **55** of the annular member **54**. If one or more of the fan blade track panels **74** are damaged a new cassette **80** with fan blade track panels **74** adhesively bonded to the cassette **80** is installed.

The downstream end of the skin **76** may comprise high-density filler for ice impact protection.

The advantage of the present invention is that the cassette comprises a metal and the cassette helps contain a fan blade in the event of a fan blade becoming detached. In particular the upstream ends of the cassettes are shaped to match the annular member on the metal casing and the upstream ends of the cassettes form an annulus and thus protect the annular mem-

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ber of the metal casing in the event of a fan blade becoming detached. The use of the cassette enables the annular member on the metal casing to be made thinner and also enables the metal casing to be made thinner. In addition the metal casing may be made from lower strength material, for example aluminium. The retention members protect the metal casing during fan blade root impact. The upstream ends of the retention members may protect the second annular member on the metal casing. The reduction in thickness of the metal casing and/or the lower strength metal enables the weight of the metal casing to be reduced. Alternatively, the cassettes may be used in conjunction with a metal casing to provide improved fan blade containment capability. Thus the cassettes comprise relatively high strength materials, eg high strength metals, and are used to protect relatively low strength materials, eg low strength metals, casings and annular members to reduce weight. Alternatively the cassettes of relatively high strength materials, eg high strength metals are used to protect relatively high strength material, eg high strength metals, casings and annular members to improve fan blade containment capability.

FIG. 6 shows an alternative arrangement for the upstream end **106** of a retention member **104**. In this arrangement the upstream end **106** comprises at least one axially extending tang **107** and a number of bent tangs **109**. The bent tangs **109** are firstly bent radially outwardly and then bent radially inwardly. These tangs **107** and **109** hold the retention members **104** on the second annular member **100**. In this arrangement the upstream end **106** of the retention member **104** may not provide as good a protection for the second annular member **100**.

FIG. 7 shows an alternative fan blade containment assembly **38B**, which is similar to that shown in FIGS. 2 to 5 and like parts are denoted by like numerals. This embodiment differs in that at least one ice impact panel **120** is secured in each cassette **80**, axially downstream of the fan blade track liner panels **74**. Each ice impact panel **120** is secured to the base portion **88** of the respective cassette **80** by fastening means **120**, for example one or more radially extending screws **122** extending through countersunk apertures **124** in the base portion **88** of the cassettes **80** and threaded apertures **126** in the ice impact panel. Alternatively each ice impact panel **120** may be secured to the base portion **88** of the respective cassette **80** by adhesive bonding or each ice impact panel **120** may be secured to the base portion of the respective cassette **80** by a combination of screws and adhesive bonding. The ice impact panels **120** comprise a graphite reinforced plastic box, which may be filled with high-density filler, for example an epoxy paste filler.

FIG. 8 shows an alternative fan blade containment assembly **38C**, which is similar to that shown in FIGS. 2 to 5 and like parts are denoted by like numerals. This embodiment differs in that at least one ice impact panel **130** is secured in each cassette **80**, axially downstream of the fan blade track liner panels **74**. Each ice impact panel **130** is secured to the base portion **88** of the respective cassette **80** by adhesive bonding. The ice impact panels **130** comprise an abradable liner filled with high-density filler, for example an epoxy paste filler. In this embodiment the base portion **88** of each cassette **80** has a stepped portion **132** axially upstream of the downstream end **86** of the cassette **80** and the ice impact panel **130** is secured to the stepped portion **132** of the base portion **88** of the cassette **80**. The stepped portion **132** is arranged radially inwardly of the base portion **88** and radially outwardly of the axially extending member **96**.

FIG. 9 shows an alternative fan blade containment assembly **38D**, which is similar to that shown in FIGS. 2 to 5 and

like parts are denoted by like numerals. This embodiment differs in that a single cylindrical annular retention member **140** is provided rather than a number of part annular members. The axially downstream end of the annular retention member **140** has a radially extending flange **142**, which is securable between the downstream flange **44** at the downstream end of the metal casing **40** and the flange **52** at the upstream end of the rear portion **50** of the fan casing **26** by fastening means **146**. The fastening means **146** comprises a number of bolts **148** which extend through aligned apertures **152**, **154** and **156** in the downstream flange **44**, the flange **52** and the flange **142** and corresponding nuts **150**. The annular retention member **140** locates on inner cylindrical surfaces **158** and **160** on the inner surface **40A** of the metal casing **40**. The axially extending members **96** at the downstream ends of the cassettes **80** are secured to the upstream end of the annular retention member **140** in the same manner as previously, e.g. using screws and anchor nuts. The second annular member **100** only comprises a first portion **101** which extends in a radially inwardly direction from the metal casing **40**.

Preferably the casing comprises a metal, more preferably the casing comprises titanium, aluminium, an aluminium alloy, a titanium alloy, magnesium or magnesium alloy. Alternatively the casing comprises a composite material comprising reinforcing fibres in a matrix material. The composite material may for example comprise an organic matrix composite or a metal matrix composite. For example carbon fibres, or glass fibres, in a resin matrix, eg epoxy or silicon carbide fibres, or silicon nitride fibres, in an aluminium matrix.

It may be possible to provide a strong fibrous material wound around the metal casing.

Although the present invention has been described with reference to a fan casing and fan blades the invention may be applicable to a compressor casing and the compressor blades or to a turbine casing and turbine blades.

We claim:

1. A gas turbine engine rotor blade containment assembly comprising:

a generally cylindrical, or frustoconical, casing, the casing having an axis and at least one integral member, the at least one integral member having a first portion extending radially from the casing and a second portion extending axially from the first portion of the at least one integral member; and

at least one cassette comprising an upstream end, a base portion and a downstream end, the upstream end of the at least one cassette being arranged to extend radially inwardly from the base portion and in an axially downstream direction, the base portion of the at least one cassette being arranged to abut the inner surface of the casing,

one or more liner panels being arranged in the at least one cassette,

the upstream end of the at least one cassette being locatable on the second portion of the at least one integral member,

the upstream end of the at least one cassette being shaped to correspond to the shape of the first and second portions of the at least one integral member to protect the first and second portions of the at least one integral member, and

the downstream end of the at least one cassette having at least one axially extending member removably securable to the casing.

2. A gas turbine engine rotor blade containment assembly as claimed in claim **1**, wherein there is a plurality of cassettes.

3. A gas turbine engine rotor blade containment assembly as claimed in claim **2**, wherein the cassettes are arranged circumferentially around and within the casing.

4. A gas turbine engine rotor blade containment assembly as claimed in claim **2**, wherein the cassettes have axially extending edges, and the axially extending edges of adjacent cassettes overlap.

5. A gas turbine engine rotor blade containment assembly as claimed in claim **4**, wherein an axially extending edge of one of the adjacent cassettes has a tongue and an axially extending edge of another of the adjacent cassettes has a groove.

6. A gas turbine engine rotor blade containment assembly as claimed in claim **1**, wherein the upstream end of the at least one cassette is U-shape, or C-shape, in cross-section in planes containing the axis of the casing.

7. A gas turbine engine rotor blade containment assembly as claimed in claim **1**, wherein the at least one integral member is annular.

8. A gas turbine engine rotor blade containment assembly as claimed in claim **1**, wherein the casing comprises a metal.

9. A gas turbine engine rotor blade containment assembly as claimed in claim **8**, wherein the casing comprises one of titanium, aluminium, an aluminium alloy, a titanium alloy, magnesium and magnesium alloy.

10. A gas turbine engine rotor blade containment assembly as claimed in claim **1**, wherein the at least one cassette comprises a metal.

11. A gas turbine engine rotor blade containment assembly as claimed in claim **10**, wherein the at least one cassette comprises one of steel, titanium and a titanium alloy.

12. A gas turbine engine rotor blade containment assembly as claimed in claim **1**, wherein the at least one axially extending member of the at least one cassette being removably securable to the casing by at least one retention member, the at least one retention member being removably securable to the casing.

13. A gas turbine engine rotor blade containment assembly as claimed in claim **1**, wherein the casing is a fan casing and rotor blades of the containment assembly are fan blades.

14. A gas turbine engine rotor blade containment assembly comprising:

a generally cylindrical, or frustoconical, casing, the casing having an axis and at least one member, the at least one member having a first portion extending radially from the casing and a second portion extending axially from the first portion of the at least one member; and

a plurality of cassettes, each having a first end and a second end, one or more liner panels being arranged in the cassettes, the first end of the cassettes being locatable on the second portion of the at least one member, the first end of the cassettes being shaped to correspond to the shape of the first and second portions of the at least one member to protect the first and second portions of the at least one member, the second end of the cassettes having at least one axially extending member removably securable to the casing, wherein the cassettes have axially extending edges, and

the axially extending edges of adjacent cassettes overlap.

15. A gas turbine engine rotor blade containment assembly comprising:

a generally cylindrical, or frustoconical, casing, the casing having an axis and at least one member, the at least one member having a first portion extending radially from the casing and a second portion extending axially from the first portion of the at least one member; and

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a plurality of cassettes, each having a first end and a second end, one or more liner panels being arranged in the cassettes, the first end of the cassettes being locatable on the second portion of the at least one member, the first end of the cassettes being shaped to correspond to the shape of the first and second portions of the at least one member to protect the first and second portions of the at least one member, the second end of the cassettes having at least one axially extending member removably securable to the casing, wherein

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the cassettes have axially extending edges, the axially extending edges of adjacent cassettes overlap, and an axially extending edge of one of the adjacent cassettes has a tongue and an axially extending edge of another of the adjacent cassettes has a groove.

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