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(54) **UAV HAVING HERMETICALLY SEALED MODULARIZED COMPARTMENTS AND FLUID DRAIN PORTS**

(71) Applicant: **AEROVIRONMENT, INC.**, Simi Valley, CA (US)

(72) Inventors: **Pavel Belik**, Simi Valley, CA (US); **Thomas Szarek**, Oak Park, CA (US); **Gabriel E. Torres**, Simi Valley, CA (US); **Vernon P. Fraye**, Simi Valley, CA (US)

(73) Assignee: **AeroVironment, Inc.**, Arlington, VA (US)

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**B64C 39/02** (2006.01)  
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CPC ..... **B64C 35/008** (2013.01); **B60F 5/00** (2013.01); **B64C 25/52** (2013.01); **B64C 39/024** (2013.01); **B64D 9/00** (2013.01); **B64C 2201/042** (2013.01)

(58) **Field of Classification Search**

CPC ..... B60F 5/00; B64C 35/008; B64C 35/024; B64C 25/52

See application file for complete search history.

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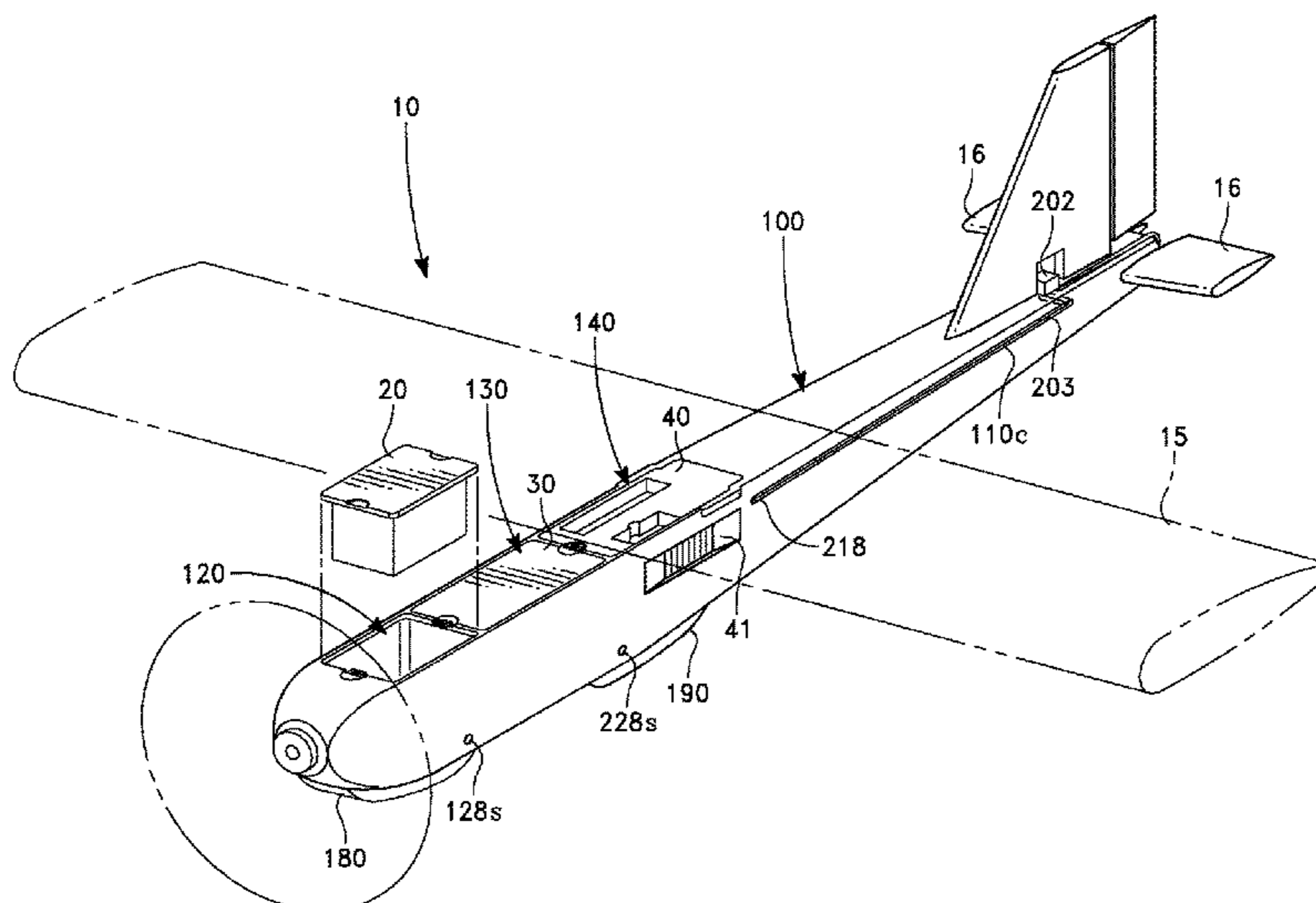
*Primary Examiner* — Justin M Benedik

(74) *Attorney, Agent, or Firm* — Christopher R. Balzan, Esq.; Eric J. Aagaard, Esq.

(57) **ABSTRACT**

In one possible embodiment, an amphibious unmanned aerial vehicle is provided, which includes a fuselage comprised of a buoyant material. Separators within the fuselage form separate compartments within the fuselage. Mounts associated with the compartments for securing waterproof aircraft components within the fuselage. The compartments each have drainage openings in the fuselage extending from the interior of the fuselage to the exterior of the fuselage.

**20 Claims, 3 Drawing Sheets**



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continuation of application No. 13/730,825, filed on Dec. 28, 2012, now Pat. No. 9,365,088, which is a continuation of application No. PCT/US2011/042285, filed on Jun. 29, 2011.

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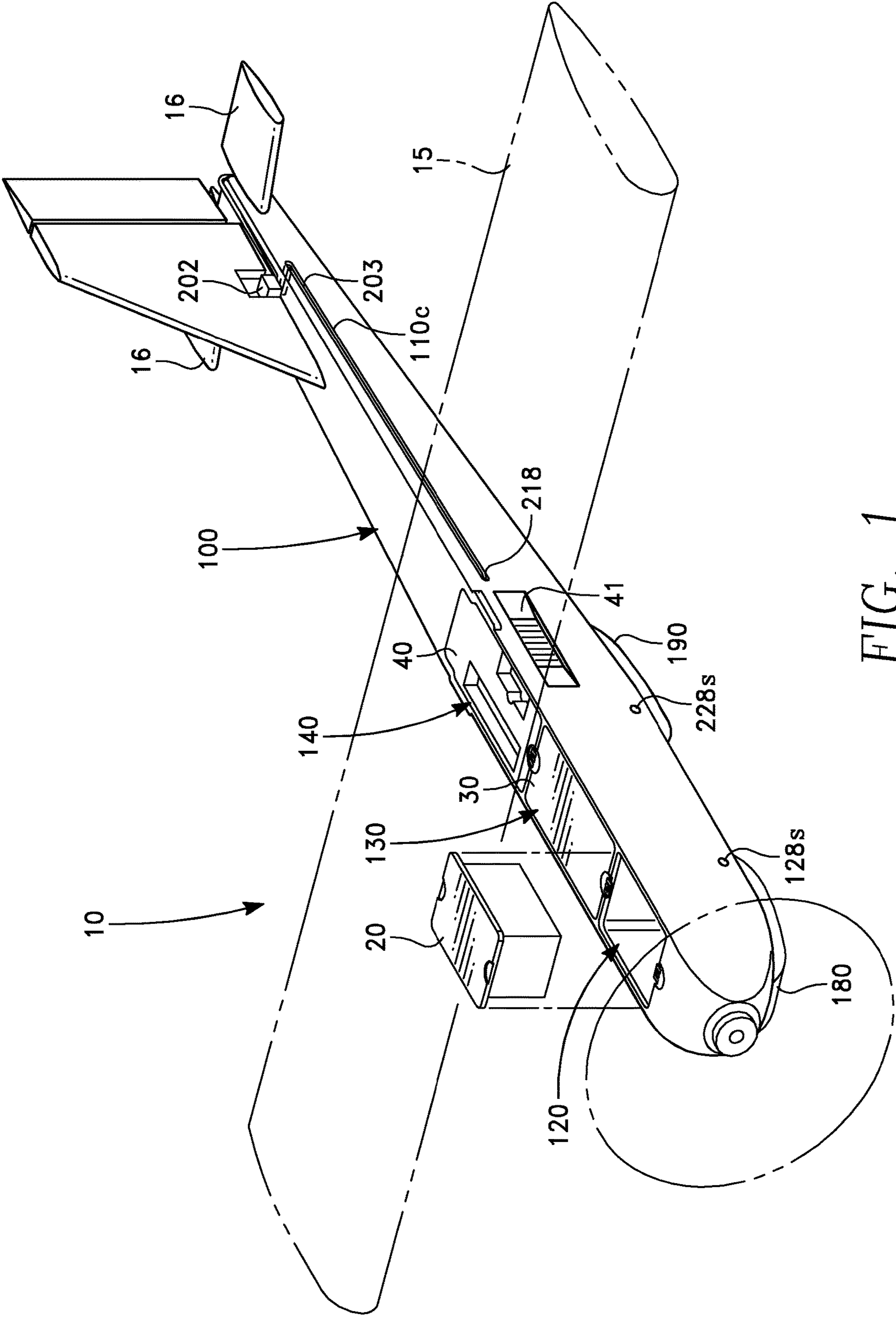


FIG. 1

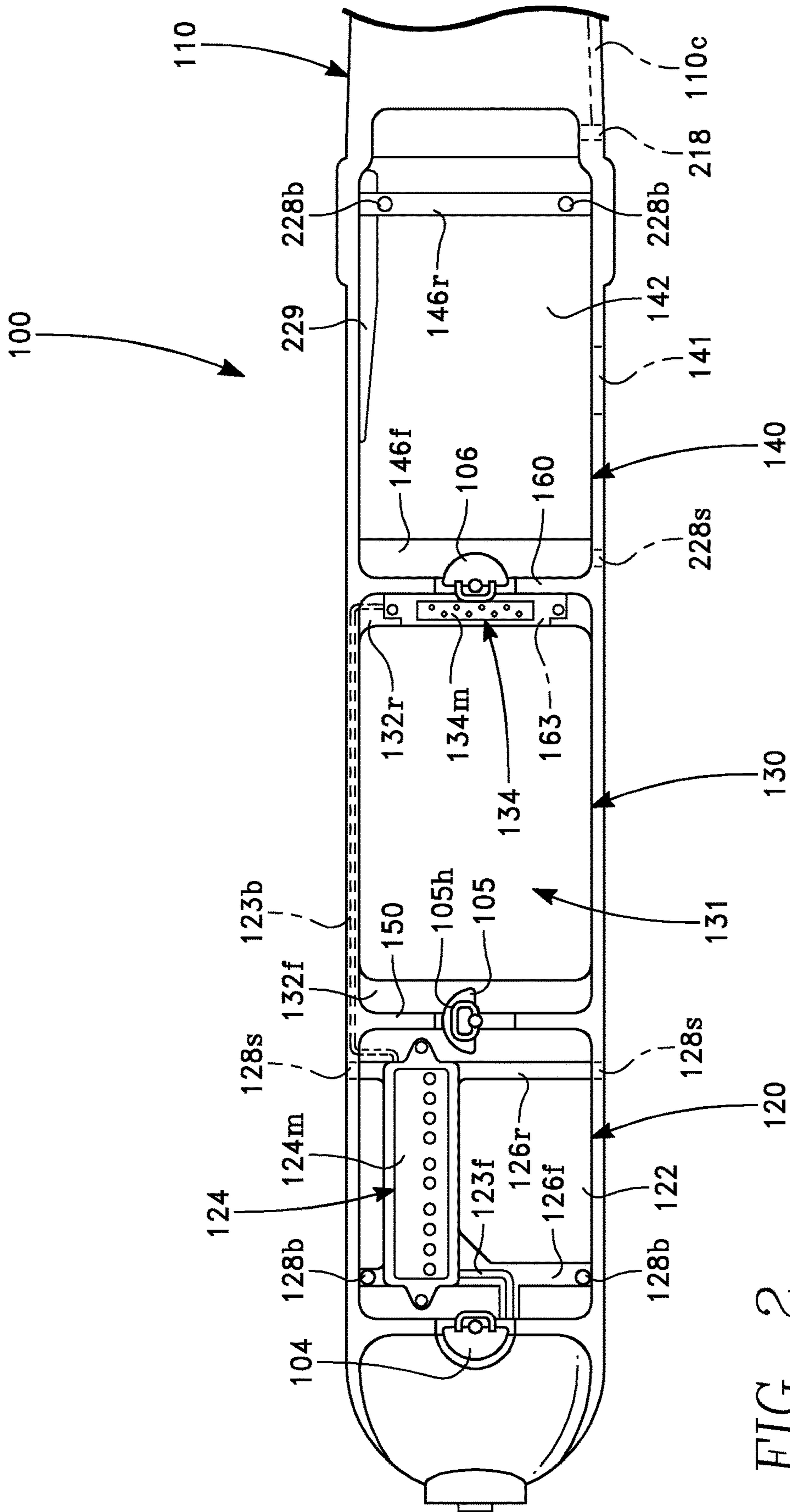


FIG. 2

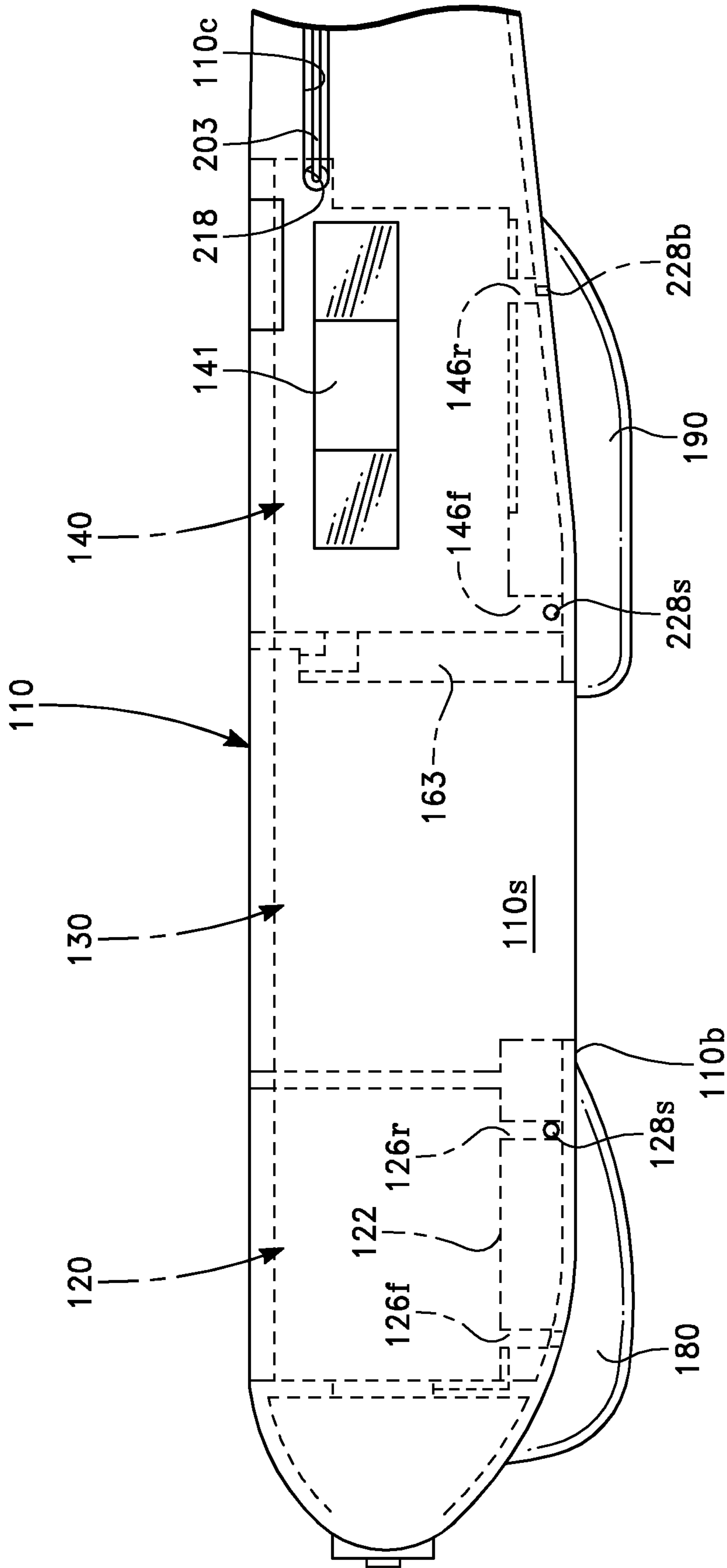


FIG. 3

**UAV HAVING HERMETICALLY SEALED  
MODULARIZED COMPARTMENTS AND  
FLUID DRAIN PORTS**

CROSS REFERENCE TO RELATED  
APPLICATIONS

The present application is a continuation of U.S. patent application Ser. No. 15/164,552, filed May 25, 2016, by Belik et al., entitled UAV HAVING HERMETICALLY SEALED MODULARIZED COMPARTMENTS AND FLUID DRAIN PORTS, which is a continuation of U.S. patent application Ser. No. 13/730,825, filed Dec. 28, 2012, by Belik et al., entitled UAV HAVING HERMETICALLY SEALED MODULARIZED COMPARTMENTS AND FLUID DRAIN PORTS, which is a continuation of PCT Application number PCT/US2011/042285, by Belik et al., entitled UAV HAVING HERMETICALLY SEALED MODULARIZED COMPARTMENTS AND FLUID DRAIN PORTS, filed 29 Jun. 2011, which are herein incorporated by reference in their entirety, which claims priority of U.S. Provisional Application No. 61/359,809, by Belik et al., filed on Jun. 29, 2010, herein incorporated by reference in its entirety.

BACKGROUND

Reducing weight and size are paramount in the design of small unmanned vehicles. Small unmanned aerial vehicles or UAVs typically are designed to be launched from, and land on dry land. Such vehicles are now being sought that can operate while being exposed to, or after being exposed to aquatic environments. For example, it may be preferred to land an unmanned aerial vehicle on water, rather than on land, either to lessen the impact of landing, or because it is more easily retrievable location. Typically, amphibious aircraft, both manned and unmanned, are able to take off and land in water.

Hand launched amphibious UAVs though are not required to take off from the water, but are required to land on dry land, or on water. Some hand launched UAVs are designed to land by skidding along, or impacting the ground, which is considerably more abrupt than water landings.

What is needed is an amphibious UAV that can withstand high impact ground landings.

SUMMARY

In one possible embodiment, an amphibious unmanned aerial vehicle is provided, which includes a fuselage comprised of a buoyant material. Separators within the fuselage form separate compartments within the fuselage. Mounts associated with the compartments for securing waterproof aircraft components within the fuselage. The compartments each have drainage openings in the fuselage extending from the interior of the fuselage to the exterior of the fuselage.

In one possible embodiment, an amphibious unmanned aerial vehicle is provided, which includes a fuselage constructed of a buoyant material and separators within the fuselage forming separate battery, payload, and avionics compartments within the fuselage. Mounts associated with the compartments are provided for securing a battery, a payload, and avionics within the fuselage. Drainage openings extend through the wall of fuselage to connect the compartments with the exterior of the unmanned aerial vehicle.

In one possible embodiment, an amphibious unmanned aerial vehicle is provided including a fuselage constructed of a buoyant material with separators in the fuselage forming separate battery, payload, and avionics compartments within the fuselage. The battery compartment and the avionics compartment have a drainage channels having a weep holes through the fuselage. The payload compartment has an open bottom. Mounting surfaces within the battery, payload, and avionics compartments are provided for mounting a battery module, a payload module, and an avionics module within the fuselage.

BRIEF DESCRIPTION OF THE DRAWINGS

The features and advantages of the present invention will be better understood with regard to the following description, appended claims, and accompanying drawings where:

FIG. 1 shows a simplified perspective view of an amphibious unmanned aerial vehicle.

FIG. 2 shows a simplified top view of the fuselage of the amphibious unmanned aerial vehicle of FIG. 1.

FIG. 3 shows a simplified side view of the fuselage of an amphibious unmanned aerial vehicle of FIG. 1.

DESCRIPTION

FIG. 1 shows a simplified perspective view of an amphibious unmanned aerial vehicle or UAV 10. The UAV 10 has a fuselage 100 of the amphibious unmanned aerial vehicle 10 has modularized compartments 120, 130, and 140 to contain modular components or modules, such as a battery module 20, a payload module 30, and avionics electronics module 40. In various embodiments, the wings 15 and/or 16 may be constructed of multiple pieces, which may separate, and/or “break away” or separate from the fuselage 100 during landings.

FIG. 2 shows a simplified top view of the fuselage 100 of the amphibious unmanned aerial vehicle 10 of FIG. 1. The walls 110 of the fuselage 100 are composed of a buoyant material so that the fuselage 100 will float without wings (not shown) attached when the fuselage is fully loaded with components, such as the battery 20, the payload 30, and the avionics electronics 40, shown in FIG. 1, and other aircraft parts and components. For example, the walls 110 may have a molded foam core sealed with a waterproof skin, though this is not required. The walls 110 may be a single continuous wall or multiple wall sections, or the like.

In this embodiment, the fuselage is separated into three compartments, a forward battery compartment 120, a central payload compartment 130, and a rear avionics compartment 140. The forward battery compartment 120 is separated from the central payload compartment 130 by separator wall 150. The central payload compartment 130 is separated from the rear avionics compartment 140 by separator wall 160. In the embodiment shown, tabs 104, 105, and 106 are employed as a means to secure the components (not shown) in the compartments 120, 130, and 140. The tab 105 may be rotated by hand using the pivotable handle 105h to allow installation of a battery (not shown) and then rotated back to the position shown to lock the battery in the forward battery compartment 120. Other securing mechanisms may be used instead or in addition to the rotatable tabs 104, 105, and 106.

The battery compartment 120 has mounting surfaces 122 which support a battery (not show). In this embodiment, a connector 124, which may be a surface mount connector or the like, is generally flush with the mounting surfaces 122. Channels 126f and 126r are recessed below the mounting

surfaces **122**. Drainage openings such as weep holes **128b** in the channel **126f** extend through the bottom wall **110b** of the fuselage **100**. Weep holes **128s** (shown in FIGS. 1-3) in the channel **126r** extend through the side wall **110s** of the fuselage **100**.

The mating surface **124m** of the connector **124** is located above the channels **126f** and **126r** so that the mating surface is not submerged in water when the battery **20** (FIG. 1) is connected/disconnected, if the fuselage is out of the water. Wiring **123f** and **123b** may be routed in the channels **126f** and **126b**, respectively, and recessed and/or embedded through the fuselage **100** to provide power to the motor (not shown) and avionics electronics module **40** and/or a payload module **30**.

The central payload compartment **130** has front and rear mounting surfaces **132f** and **132r** which support a payload, such as a camera assembly (not shown). The payload module **30** may contain the imaging, sensing, or other passive, active, non-lethal, or lethal payload devices. In this embodiment, a connector **134**, which may be a surface mount connector or the like, is generally flush with the mounting surface **132r**. The mounting surface **132r** may form an enclosure **163** to contain the connector **134** and associated wiring. The enclosure may form a lower part of the separator wall **160**. Weep holes **228s** (shown in FIGS. 1-3) may extend from inside the enclosure **163** through the side wall **110s** to allow water to exit the enclosure **163**. In this embodiment, the central compartment **130** has a large opening **131** in the bottom so that a camera can be utilized, such as by viewing downward or by lowering it into the airstream through the large opening **131**. The large opening **131** also allows drainage of fluid from the central compartment **130**.

In various embodiments, the mating surface **134m** of the connector **134** may be located high above the opening **131**, on the top of the enclosure **163**, so that the mating surface is not submerged in water when the payload **30** (FIG. 1) is connected/disconnected, even if the fuselage is not completely out of the water.

The rear avionics compartment **140** has a mounting surface **142** in a bottom of the avionics compartment **140**. The mounting surface **142** has a forward channel **146f** and a rear channel **146r**. Channels **146f** and **146r** are recessed below the mounting surface **142**. Drainage openings such as weep hole **228s** (shown in FIGS. 1-3) in channel **146f** extends through the side wall **110s** of the fuselage **100**. Weep holes **228b** (shown in FIGS. 2 and 3) in the channel **146r** extend through the bottom wall **110b** of the fuselage **100**. A sloping recess **229** in the mounting surface **142** drains water off the mounting surface **142** and into the channel **146r**.

The embodiment shown in FIG. 3 has an opening **141** in the side wall **110s** of fuselage **100**, to expose a heat sink **41** (FIG. 1) and allow heat generated by the avionics electronics **40** (FIG. 1) to be released.

FIG. 3 shows a simplified side view of the fuselage **100** of an amphibious unmanned aerial vehicle **10** of FIG. 1. In this embodiment, optional skid pads **180** and **190** are secured to the bottom wall **110b** of the fuselage **100**. The skid pads **180** and **190** are used in this embodiment for landing on hard surfaces. The skid pad **180** may be located directly below the forward compartment **120** and may be fabricated of a durable shock absorbing material of sufficient thickness and density to further protect from impact the component within the compartment **120**, such as the battery **20** (FIG. 1). Similarly, the skid pad **190** may be located directly below the rear compartment **140**, and may be fabricated of a durable shock absorbing material of sufficient thickness and

density to further protect from impact a component within the compartment **140**, such as the avionics electronics **40** (FIG. 1).

Weep holes **128s** extend through the side wall **110s** of the fuselage **100**. The weep hole **128s** extends through the side wall **110s** and into rear channel **126r** of the battery compartment **120**. The weep hole **228s** extends through the side wall **110s** and into the enclosure **163** of the central payload compartment **130**.

The fluid drainage openings may be weep holes, fluid drainage ports, or the like.

Various embodiments provide a fuselage **100** for a UAV which can land both on water and rugged terrain. Instead of sealing the entire aircraft from water intrusion, various embodiments achieve the ability to land on water by having just the individual electrical and electronic components, i.e. battery, payload, avionics electronics, and associated connectors and wiring, hermetically sealed.

This allows the rest of the aircraft to remain buoyant and any water in the aircraft draining by a set of fluid drain ports when the UAV is retrieved from the water. In this manner, the protection of the electrical and electronic components is not dependant on maintaining the integrity of the fuselage **100** or the exterior walls **110**, which is likely to be damaged during landings on hard and/or rugged surfaces (typical of land landings).

This also allows the volume within the aircraft that needs to be waterproof to be minimized, thus reducing weight and overall system complexity.

Further, the aircraft's ability to land on hard surfaces or rugged terrain without damage to the electrical and electronic components is achieved not just by having these components encased in modularized compartments **120**, **130**, and **140**, but also by allowing the walls **110** of the compartments **120**, **130**, and **140** to be partially compromised without necessarily causing failure of the UAV. The walls **110** create an impact zone around the electrical and electronic components within the compartments **120**, **130**, and **140**, and the separators inhibit the components **20**, **30**, and **40** from impacting each other. Optionally, in some embodiments, the walls **110** and mounts **122**, **132f**, **132r**, and **142** are such that the components **20**, **30**, and **40** (FIG. 1) are recessed from the walls **110** and/or their respective separators **150** and **160** (FIG. 2). Additional shock absorbing material (not shown) may be added within the compartments **120**, **130**, or **140** to further reduce any chance of damage to the components **20**, **30**, or **40** from impacts.

As illustrated in FIGS. 1 and 2, the fuselage **100** may contain an optional exterior channel **110c** in the side **110s** of the fuselage **100**, extending rearward from a hole **218** in the side wall **110s** at the avionics compartment **140**, to the tail section of the aircraft **10**. Wiring **203** extends through the hole **218** and along the exterior channel **110c** to connect the avionics component **40** to an actuator assembly **202** for actuating the control surfaces in the tail of the aircraft **10**. The exterior channel **110c** allows the wiring to be easily accessed for inspection, repair, and replacement.

It is worthy to note that any reference to "one embodiment" or "an embodiment" means that a particular feature, structure, or characteristic described in connection with the embodiment may be included in an embodiment, if desired. The appearances of the phrase "in one embodiment" in various places in the specification are not necessarily all referring to the same embodiment.

The illustrations and examples provided herein are for explanatory purposes and are not intended to limit the scope of the appended claims. This disclosure is to be considered



## 5

an exemplification of the principles of the invention and is not intended to limit the spirit and scope of the invention and/or claims of the embodiment illustrated.

Those skilled in the art will make modifications to the invention for particular applications of the invention.

The discussion included in this patent is intended to serve as a basic description. The reader should be aware that the specific discussion may not explicitly describe all embodiments possible and alternatives are implicit. Also, this discussion may not fully explain the generic nature of the invention and may not explicitly show how each feature or element can actually be representative or equivalent elements. Again, these are implicitly included in this disclosure. Where the invention is described in device-oriented terminology, each element of the device implicitly performs a function. It should also be understood that a variety of changes may be made without departing from the essence of the invention. Such changes are also implicitly included in the description. These changes still fall within the scope of this invention.

Further, each of the various elements of the invention and claims may also be achieved in a variety of manners. This disclosure should be understood to encompass each such variation, be it a variation of any apparatus embodiment, a method embodiment, or even merely a variation of any element of these. Particularly, it should be understood that as the disclosure relates to elements of the invention, the words for each element may be expressed by equivalent apparatus terms even if only the function or result is the same. Such equivalent, broader, or even more generic terms should be considered to be encompassed in the description of each element or action. Such terms can be substituted where desired to make explicit the implicitly broad coverage to which this invention is entitled. It should be understood that all actions may be expressed as a means for taking that action or as an element which causes that action. Similarly, each physical element disclosed should be understood to encompass a disclosure of the action which that physical element facilitates. Such changes and alternative terms are to be understood to be explicitly included in the description.

Having described this invention in connection with a number of embodiments, modification will now certainly suggest itself to those skilled in the art. The example embodiments herein are not intended to be limiting, various configurations and combinations of features are possible. As such, the invention is not limited to the disclosed embodiments, except as required by the appended claims.

What is claimed is:

1. An amphibious unmanned aerial vehicle comprising:
  - a) a fuselage comprised of a buoyant material walls;
  - b) separators within the fuselage forming separate compartments within the fuselage;
  - c) mounts associated with the compartments for securing waterproof aircraft components within the fuselage; and
  - d) wherein the compartments are constructed to individually surround a battery, a payload, and avionics electronics, but wherein the compartments are not sealed from water intrusion, while the battery, the payload, the avionics electronics, and associated connectors and wiring are hermetically sealed such that the amphibious unmanned aerial vehicle is capable of landing on land and on water without damage to electrical and electronic components.
2. The amphibious unmanned aerial vehicle of claim 1, wherein the compartments are modularized compartments

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comprising openings to allow loading of a battery module, a payload module, and an avionics electronics module therein.

3. The amphibious unmanned aerial vehicle of claim 2 comprising a securing mechanism to retain the battery module, the payload module, and the avionics electronics module within their respective modularized compartments.

4. The amphibious unmanned aerial vehicle of claim 3, wherein the securing mechanism comprises hand rotatable tabs.

5. The amphibious unmanned aerial vehicle of claim 4, wherein the hand rotatable tabs comprise a pivotable handle.

6. The amphibious unmanned aerial vehicle of claim 1, wherein the compartments each have drainage openings in the fuselage extending from the interior of the fuselage to the exterior of the fuselage.

7. The amphibious unmanned aerial vehicle of claim 6, wherein the drainage openings in some of the compartments comprise weep holes through a wall of the fuselage.

8. The amphibious unmanned aerial vehicle of claim 1, wherein at least one of the compartments comprises a drainage channel at a bottom thereof, and wherein the drainage channel comprises a drainage opening associated therewith.

9. The amphibious unmanned aerial vehicle of claim 8, wherein the drainage opening comprises a weep hole.

10. The amphibious unmanned aerial vehicle of claim 8, wherein the drainage opening comprises a plurality of weep holes.

11. The amphibious unmanned aerial vehicle of claim 1, comprising a skid pad on a bottom of the fuselage.

12. The amphibious unmanned aerial vehicle of claim 1 comprising drainage openings extending through the separators within the fuselage.

13. The amphibious unmanned aerial vehicle of claim 1 comprising securing mechanisms at a top of the fuselage adjacent to openings in the compartments to secure the waterproof aircraft components within the compartments.

14. The amphibious unmanned aerial vehicle of claim 1, wherein at least some of the mounts within the compartments comprise integrated connectors for electrically connecting a corresponding aircraft component upon mounting of the corresponding aircraft component within a respective compartment.

15. The amphibious unmanned aerial vehicle of claim 1, wherein the compartments comprise a battery compartment comprising a surface mount connector flush with a mounting surface of the mounts within the battery compartment.

16. The amphibious unmanned aerial vehicle of claim 1, wherein the compartments comprise a payload compartment comprising a surface mount connector flush with a mounting surface of the mounts within the payload compartment.

17. The amphibious unmanned aerial vehicle of claim 1, wherein the compartments comprise a payload compartment comprising an open bottom.

18. The amphibious unmanned aerial vehicle of claim 17, wherein there is no fuselage wall below the payload compartment.

19. The amphibious unmanned aerial vehicle of claim 1, wherein the compartments comprise an avionics compartment comprising a heat sink opening.

20. The amphibious unmanned aerial vehicle of claim 1 comprising:

- a) the separators within the fuselage forming separate battery, payload, and avionics compartments within the fuselage to protect the battery, the payload, and the avionics electronics;

- b) the mounts associated with the compartments being for securing the battery, the payload, and the avionics electronics within the fuselage; and
- c) drainage openings extending through the wall of fuselage to connect the compartments with an exterior of the unmanned aerial vehicle.

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